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NATIONAL ADVISORY COMMITTEE  
FOR AERONAUTICS

REPORT No. 315

AERODYNAMIC CHARACTERISTICS OF AIRFOILS—VI

By

NATIONAL ADVISORY COMMITTEE  
FOR AERONAUTICS

94-15791



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## AERONAUTICAL SYMBOLS

### 1. FUNDAMENTAL AND DERIVED UNITS

	Symbol	Metric		English	
		Unit	Symbol	Unit	Symbol
Length-----	$l$	meter-----	m	foot (or mile)-----	ft. (or mi.)
Time-----	$t$	second-----	sec	second (or hour)-----	sec. (or hr.)
Force-----	$F$	weight of one kilogram-----	kg	weight of one pound-----	lb.
Power-----	$P$	kg/m/sec-----		horsepower-----	HP.
Speed-----		km/hr-----		mi./hr-----	M. P. H.
		m/sec-----		ft./sec-----	f. p. s.

### 2. GENERAL SYMBOLS, ETC.

$W$ , Weight, $=mg$	$mk^2$ , Moment of inertia (indicate axis of the radius of gyration, $k$ , by proper subscript).
$g$ , Standard acceleration of gravity $=9.80665$ m/sec. <sup>2</sup> $=32.1740$ ft./sec. <sup>2</sup>	$S$ , Area.
$m$ , Mass, $=\frac{W}{g}$	$S_w$ , Wing area, etc.
$\rho$ , Density (mass per unit volume).	$G$ , Gap.
Standard density of dry air, $0.12497$ (kg-m <sup>-4</sup> sec. <sup>2</sup> ) at $15^\circ$ C and $760$ mm $=0.002378$ (lb.-ft. <sup>-4</sup> sec. <sup>2</sup> ).	$b$ , Span.
Specific weight of "standard" air, $1.2255$ kg/m <sup>3</sup> $=0.07651$ lb./ft. <sup>3</sup>	$c$ , Chord length.
	$b/c$ , Aspect ratio.
	$f$ , Distance from $c. g.$ to elevator hinge.
	$\mu$ , Coefficient of viscosity.

### 3. AERODYNAMICAL SYMBOLS

$V$ , True air speed.	$\gamma$ , Dihedral angle.
$q$ , Dynamic (or impact) pressure $=\frac{1}{2}\rho V^2$	$\frac{VL}{\mu}$ , Reynolds Number, where $l$ is a linear dimension.
$L$ , Lift, absolute coefficient $C_L=\frac{L}{qS}$	e. g., for a model airfoil 3 in. chord, 100 mi./hr. normal pressure, $0^\circ$ C: 255,000 and at $15^\circ$ C., 230,000;
$D$ , Drag, absolute coefficient $C_D=\frac{D}{qS}$	or for a model of 10 cm chord 40 m/sec, corresponding numbers are 299,000 and 270,000.
$C$ , Cross-wind force, absolute coefficient $C_C=\frac{C}{qS}$	$C_p$ , Center of pressure coefficient (ratio of distance of $C. P.$ from leading edge to chord length).
$R$ , Resultant force. (Note that these coefficients are twice as large as the old coefficients $L_C, D_C$ .)	$\beta$ , Angle of stabilizer setting with reference to lower wing, $=(i_t - i_w)$ .
$i_w$ , Angle of setting of wings (relative to thrust line).	$\alpha$ , Angle of attack.
$i_t$ , Angle of stabilizer setting with reference to thrust line.	$\epsilon$ , Angle of downwash.

# REPORT No. 315

## AERODYNAMIC CHARACTERISTICS OF AIRFOILS—VI CONTINUATION OF REPORTS Nos. 93, 124, 182, 244, and 286

By  
NATIONAL ADVISORY COMMITTEE  
FOR AERONAUTICS

39331—29—1

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NAVY BUILDING, WASHINGTON, D. C.

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(An independent Government establishment, created by act of Congress approved March 3, 1915, for the supervision and direction of the scientific study of the problems of flight. It consists of 12 members who are appointed by the President, all of whom serve as such without compensation.)

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# REPORT No. 315

## AERODYNAMIC CHARACTERISTICS OF AIRFOILS—VI

CONTINUATION OF REPORTS NOS. 93, 124, 182, 244, AND 286

By NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

### INTRODUCTION

This collection of data on airfoils has been made from the published reports of a number of the leading aerodynamic laboratories of this country and Europe.<sup>1</sup> The information which was originally expressed according to the different customs of the several laboratories is here presented in a uniform series of charts and tables suitable for the use of designing engineers and for purposes of general reference.

It is a well-known fact that the results obtained in different laboratories, because of their individual methods of testing, are not strictly comparable even if proper scale corrections for size of model and speed of test are supplied. It is, therefore, unwise to compare too closely the coefficients of two wing sections tested in different laboratories. Tests of different wing sections from the same source, however, may be relied on to give true relative values.

The absolute system of coefficients has been used, since it is thought by the National Advisory Committee for Aeronautics that this system is the one most suited for international use and yet it is one from which a desired transformation can be easily made. For this purpose a set of transformation constants is given.

Each airfoil section is given a reference number, and the test data are presented in the form of curves from which the coefficients can be read with sufficient accuracy for designing purposes. The dimensions of the profile of each section are given at various stations along the chord in per cent of the chord length, the latter also serving as the datum line. The shape of the section is also shown with reasonable accuracy in order to enable one to more clearly visualize the section under consideration, the outside of the heavy line representing the profile.

The authority for the results here presented is given as the name of the laboratory at which the experiments were conducted, as explained under abbreviations, with the size of model, wind velocity, and year of test.

### TRANSFORMATION CONSTANTS

For the convenience of those who prefer to use a system of units other than the absolute system, there is given below a table of transformation constants based on the standard condition adopted by the National Advisory Committee for Aeronautics of—

Temperature	= 15° C	= 59° F.
Pressure	= 760 mm Hg	= 29.92 in. Hg.
Humidity	= 0	
Gravity	= 9.80665 m/s <sup>2</sup>	= 32.1740 ft./sec. <sup>2</sup>

thus giving values of specific weight of air

$$W = 1.2255 \text{ kg/m}^3 = 0.07651 \text{ lb./cu. ft.}$$

<sup>1</sup> A previous collection of airfoil sections numbered 1 to 759 and Charts 1 to 20 may be found in N. A. C. A. Reports Nos. 93, 124, 182, 244, and 286.

and of density

$\rho = 0.12497 \text{ kg-m}^{-3}$  in the French engineering or kilogram, meter, second system.

Or

$\rho = 0.002378 \text{ lb.-ft.}^{-3}$  in the English or pound, foot, second system.

In absolute units .....	$P = CV^2 \rho / 2$
In $\text{kg/m}^2$ (m/s) .....	$P = .0625 CV^2$
In $\text{kg/m}^2$ (km/h) .....	$P = .004822 CV^2$
In $\text{lb./sq. ft. (ft./sec.)}$ .....	$P = .001189 CV^2$
In $\text{lb./sq. ft. (mi./hr.)}$ .....	$P = .002558 CV^2$

(Note that these constants are half as large as those used in Reports Nos. 93 and 124 and that the absolute coefficients used in this report are twice as large as the old coefficients. See Report No. 240 regarding change in absolute coefficients.)

#### INDEX

Four separate types of indexes are given—chart indexes which make it possible for a designer to select the wing section most suitable for the particular design in which he is interested; a group index which is arranged by countries and laboratories at which tests were conducted, each section also being designated by a reference number; an index of abbreviations, used on the characteristic sheets, to indicate the laboratories at which the tests were made; and an alphabetical index.

#### CHART INDEX

In order that the designer may easily pick out a wing section which is suited to the type of airplane on which he is working, four index charts are given which classify the wings according to their aerodynamic and structural properties. In the charts of this report a lower-case letter is placed adjacent to the reference number giving  $Vl$  values, so that a comparison can be made without referring to the individual drawings. In this value  $V$  represents the wind velocity in feet per second and  $l$  a linear dimension, the chord length in feet.

In chart No. 21 the minimum drag  $C_D$ , is plotted against the  $L/D$  at one-fourth the maximum lift  $C_L$ . This chart should be used in choosing a wing section for a high-speed airplane, the wing sections being more suited for this use the farther they are from the lower left-hand corner.

In chart No. 22 the mean spar depth is plotted against the maximum lift  $C_L$ , in order to show the possible strength and lightness of the wing structure. The higher the maximum lift coefficient is, the smaller will be the wing area and the lighter the structural weight, and in the same way the greater the depth of the spars the lighter will be their weight, so that the sections the greatest distance from the lower left-hand corner will give the lightest and strongest wings. The "mean spar depth" is obtained by assuming the spars to be located, respectively, at 15 and 60 per cent of the chord, and by dividing the sum of their thicknesses, in per cent of chord length at these points, by 2.

In chart No. 23 the maximum  $L/D$  is plotted against the maximum lift  $C_L$ , which is of use in choosing the wing section for a slow and efficient airplane. In the same way as before the sections farthest from the lower left-hand corner are the best for this purpose.

In chart No. 24 the  $L/D$  at two-thirds the maximum lift  $C_L$ , is plotted against the maximum lift  $C_L$ . This chart can be used for choosing a section that will give an efficient climb or a long range at cruising speed. The best sections for this purpose will be farthest from the lower left-hand corner of the chart.

## CHART INDEX

Chart No. 21. Minimum drag $C_D$ , plotted against $L/D$ at one-fourth the maximum lift $C_L$ .....	Page 415
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Chart No. 23. Maximum $L/D$ plotted against maximum lift $C_L$ .....	417
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## INDEX OF ABBREVIATIONS

Name of laboratory at which tests were made	Abbreviations used on figures
Langley Memorial Aeronautical Laboratory of the National Advisory Committee for Aeronautics, U. S. A.	L. M. A. L.
Washington Navy Yard, U. S. A. ....	W. N. Y.
Engineering Division, McCook Field, U. S. A. ....	McC. F.
Ergebnisse der Aerodynamischen Versuchsanstalt zu Göttingen, Germany .....	Göttingen.
Service Technique de l'Aéronautique, France .....	S. T. Aé.
Laboratoire Aerotechnique de Rhode St. Genese-Bruxelles, Belgium .....	Rhode St. Genese.
Istituto Sperimentale Aeronautico, Italy .....	I. S. A.



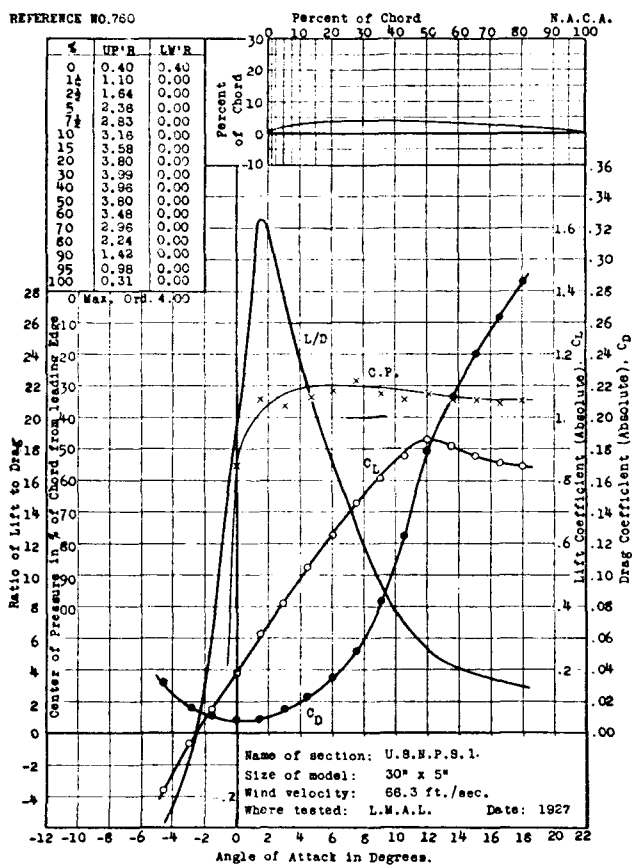
# GROUP INDEX

Airfoil	Wind tunnel where tested	Report reference number	Airfoil	Wind tunnel where tested	Report reference number
UNITED STATES			FRANCE—continued		
U. S. N. P. S. 1	L. M. A. L.	760	St. Cyr 162 (Royer)	S. T. Aé. Lab.	810
U. S. N. P. S. 2	do.	761	St. Cyr 170 (Royer)	do.	811
U. S. N. P. S. 3	do.	762	St. Cyr 174 (Royer)	do.	812
U. S. N. P. S. 4	do.	763	St. Cyr 233 (Bartel 37-IIb)	do.	813
U. S. N. P. S. 5	do.	764	St. Cyr 235 (Bartel 37-Ic)	do.	814
U. S. N. P. S. 6	do.	765	St. Cyr 237 (Bartel 15-Ic)	do.	815
Martin M-I	W. N. Y.	766	St. Cyr 239 (Bartel 37-IIa)	do.	816
Martin M-II	do.	767	St. Cyr 240 (Bartel 37-IIb)	do.	817
Martin M-III	do.	768	St. Cyr 242 (Bartel 57-IIc)	do.	818
R-3 (Root section)	McC. F.	769	St. Cyr 243 (Bartel 37-IIIc)	do.	819
U. S. A. 45M	do.	770			
GERMANY			BELGIUM		
Göttingen 359	Göttingen	771	Rhode St. Genese 1	Rhode St. Genese	820
Göttingen 361	do.	772	Rhode St. Genese 2	do.	821
Göttingen 362	do.	773	Rhode St. Genese 16	do.	822
Göttingen 368	do.	774	Rhode St. Genese 17	do.	823
Göttingen 369	do.	775	Rhode St. Genese 19	do.	824
Göttingen 371	do.	776	Rhode St. Genese 26	do.	825
Göttingen 372	do.	777	Rhode St. Genese 29	do.	826
Göttingen 373	do.	778	Rhode St. Genese 31	do.	827
Göttingen 374	do.	779	Rhode St. Genese 33	do.	828
Göttingen 375	do.	780	Rhode St. Genese 35	do.	829
Göttingen 377	do.	781	Rhode St. Genese 37	do.	830
Göttingen 392	do.	782			
Göttingen 397	do.	783	ITALY		
Göttingen 399	do.	784	I. S. A. 334	I. S. A.	831
Göttingen 401	do.	785	I. S. A. 390	do.	832
Göttingen 402	do.	786	I. S. A. 472	do.	833
Göttingen 403	do.	787	I. S. A. 500	do.	834
Göttingen 408	do.	788	I. S. A. 501	do.	835
Göttingen 417	do.	789	I. S. A. 502	do.	836
Göttingen 428	do.	790	I. S. A. 507	do.	837
Göttingen 437	do.	791	I. S. A. 605	do.	838
Göttingen 438	do.	792	I. S. A. 663	do.	839
Göttingen 439	do.	793	I. S. A. 693	do.	840
Göttingen 442	do.	794	I. S. A. 695	do.	841
Göttingen 443	do.	795	I. S. A. 768	do.	842
Göttingen 444	do.	796	I. S. A. 776	do.	843
Göttingen 445	do.	797	I. S. A. 801	do.	844
			I. S. A. 803	do.	845
FRANCE			I. S. A. 804	do.	846
Eiffel 359 (Nieuport Astra)	S. T. Aé. Lab.	798	I. S. A. 805	do.	847
Eiffel 386 (S. T. Ac.)	do.	799	I. S. A. 806	do.	848
Eiffel 387 (S. T. Aé.)	do.	800	I. S. A. 807	do.	849
Eiffel 402 (Pescara)	do.	801	I. S. A. 809	do.	850
Eiffel 429 (Lachassagne)	do.	802	I. S. A. 812	do.	851
Eiffel 432 (Lachassagne)	do.	803	I. S. A. 820	do.	852
Eiffel 433a (Chalambel)	do.	804	I. S. A. 821	do.	853
Eiffel 438 (Lachassagne)	do.	805	I. S. A. 829	do.	854
St. Cyr 152 (Royer)	do.	806	I. S. A. 858	do.	855
St. Cyr 153 (Royer)	do.	807	I. S. A. 881a	do.	856
St. Cyr 156 (Royer)	do.	808	I. S. A. 881b	do.	857
St. Cyr 157 (Royer)	do.	809	I. S. A. 881c	do.	858
			I. S. A. 911	do.	859

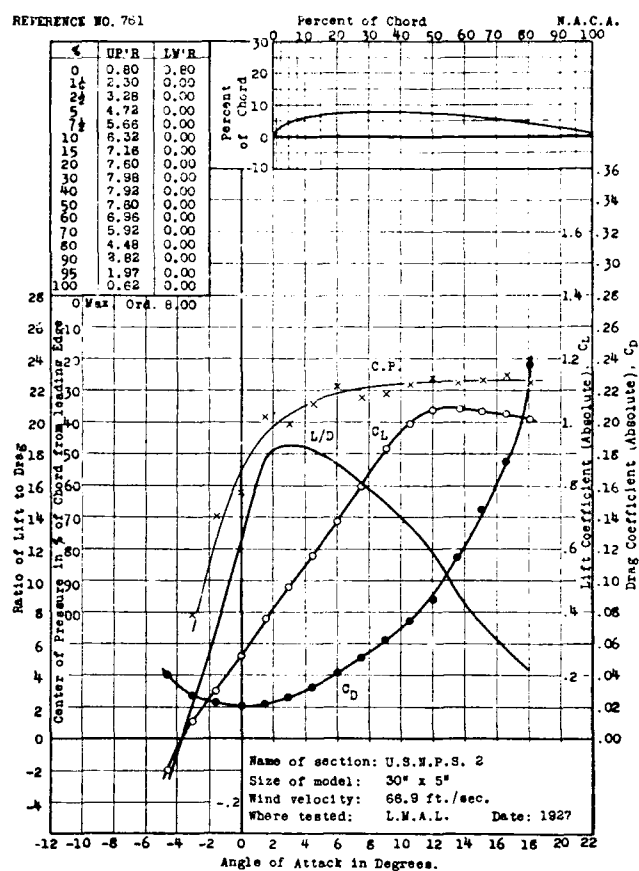
# ALPHABETICAL INDEX

Airfoil	Report reference number	Airfoil	Report reference number
Eiffel 359 (Nieuport Astra)	798	I. S. A. 804	846
Eiffel 386 (S. T. A6.)	799	I. S. A. 805	847
Eiffel 387 (S. T. A6.)	800	I. S. A. 806	848
Eiffel 402 (Pescara)	801	I. S. A. 807	849
Eiffel 429 (Lachassagne)	802	I. S. A. 809	850
Eiffel 432 (Lachassagne)	803	I. S. A. 812	851
Eiffel 433a (Chalambel)	804	I. S. A. 820	852
Eiffel 438 (Lachassagne)	805	I. S. A. 821	853
Göttingen 359	771	I. S. A. 829	854
Göttingen 361	772	I. S. A. 858	855
Göttingen 362	773	I. S. A. 881a	856
Göttingen 368	774	I. S. A. 881b	857
Göttingen 369	775	I. S. A. 881c	858
Göttingen 371	776	I. S. A. 911	859
Göttingen 372	777	Martin M-I	766
Göttingen 373	778	Martin M-II	767
Göttingen 374	779	Martin M-III	768
Göttingen 375	780	R-3 (Root Section)	769
Göttingen 377	781	Rhode St. Genese 1	820
Göttingen 392	782	Rhode St. Genese 2	821
Göttingen 397	783	Rhode St. Genese 16	822
Göttingen 399	784	Rhode St. Genese 17	823
Göttingen 401	785	Rhode St. Genese 19	824
Göttingen 402	786	Rhode St. Genese 26	825
Göttingen 403	787	Rhode St. Genese 29	826
Göttingen 408	788	Rhode St. Genese 31	827
Göttingen 417	789	Rhode St. Genese 33	828
Göttingen 428	790	Rhode St. Genese 35	829
Göttingen 437	791	Rhode St. Genese 37	830
Göttingen 438	792	St. Cyr 152 (Royer)	806
Göttingen 439	793	St. Cyr 153 (Royer)	807
Göttingen 442	794	St. Cyr 156 (Royer)	808
Göttingen 443	795	St. Cyr 157 (Royer)	809
Göttingen 444	796	St. Cyr 162 (Royer)	810
Göttingen 445	797	St. Cyr 170 (Royer)	811
I. S. A. 334	831	St. Cyr 174 (Royer)	812
I. S. A. 390	832	St. Cyr 233 (Bartel 7-Ib)	813
I. S. A. 472	833	St. Cyr 235 (Bartel 37-Ic)	814
I. S. A. 500	834	St. Cyr 237 (Bartel 15-Ic)	815
I. S. A. 501	835	St. Cyr 239 (Bartel 37-IIa)	816
I. S. A. 502	836	St. Cyr 240 (Bartel 37-IIb)	817
I. S. A. 507	837	St. Cyr 242 (Bartel 57-IIc)	818
I. S. A. 605	838	St. Cyr 243 (Bartel 37-IIc)	819
I. S. A. 663	839	U. S. A. 45M	770
I. S. A. 693	840	U. S. N. P. S. 1	760
I. S. A. 695	841	U. S. N. P. S. 2	761
I. S. A. 768	842	U. S. N. P. S. 3	762
I. S. A. 776	843	U. S. N. P. S. 4	763
I. S. A. 801	844	U. S. N. P. S. 5	764
I. S. A. 803	845	U. S. N. P. S. 6	765

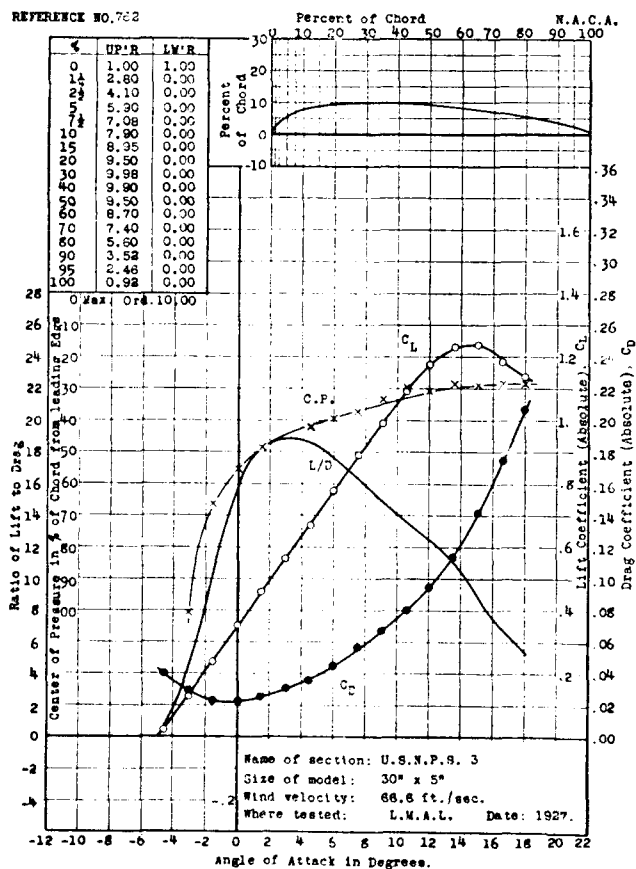
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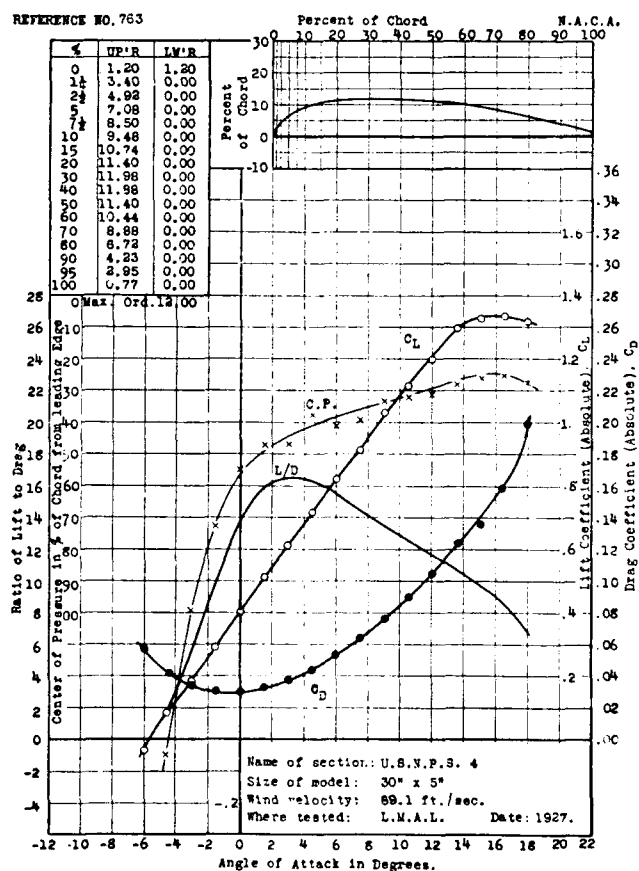
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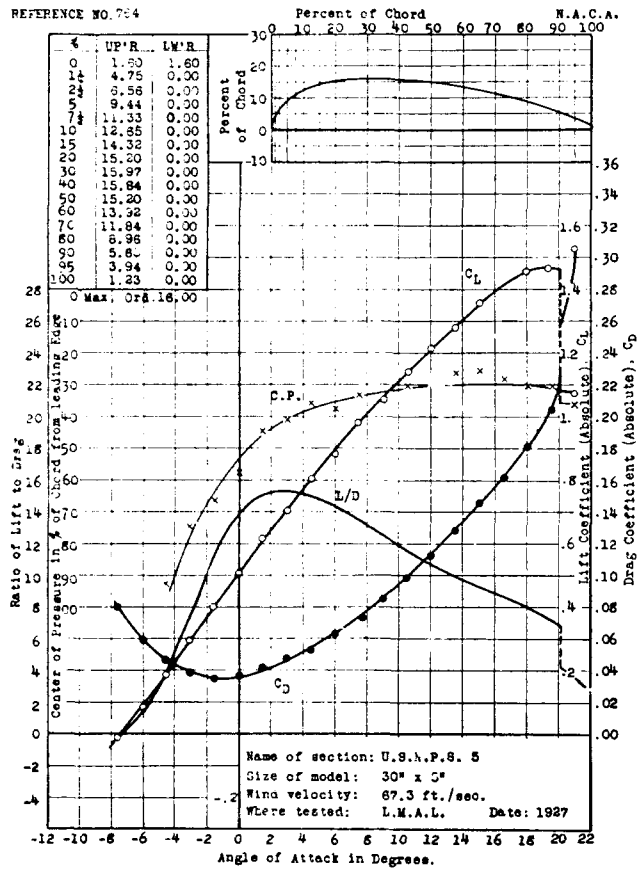
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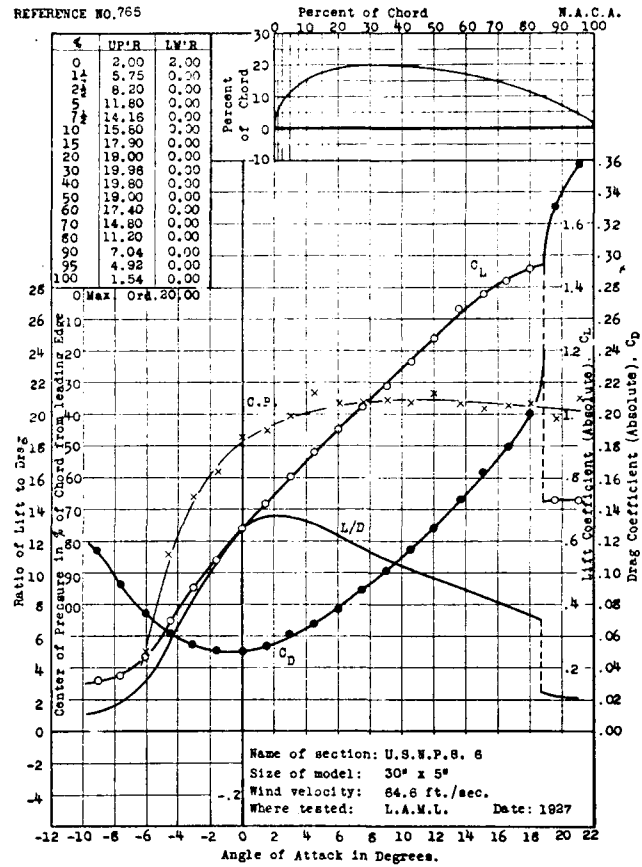
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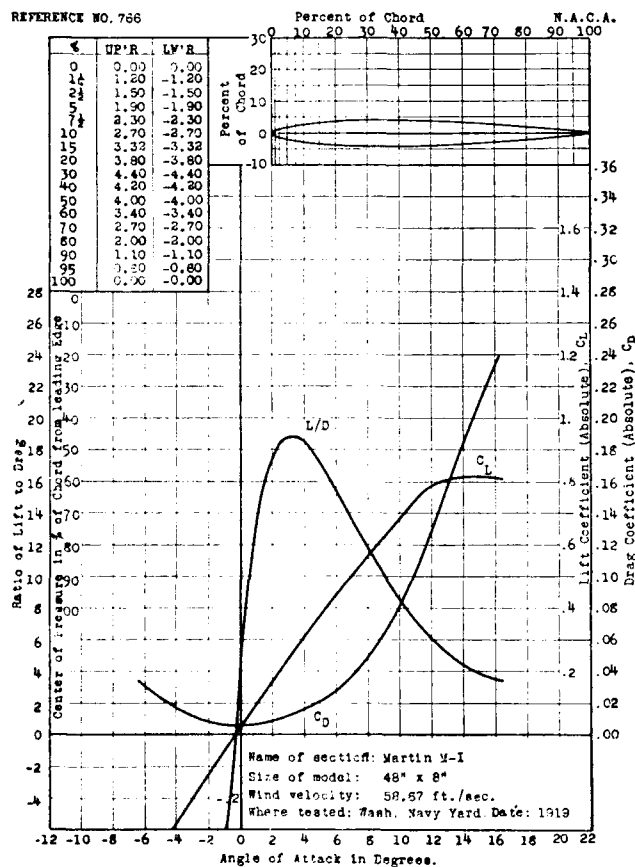
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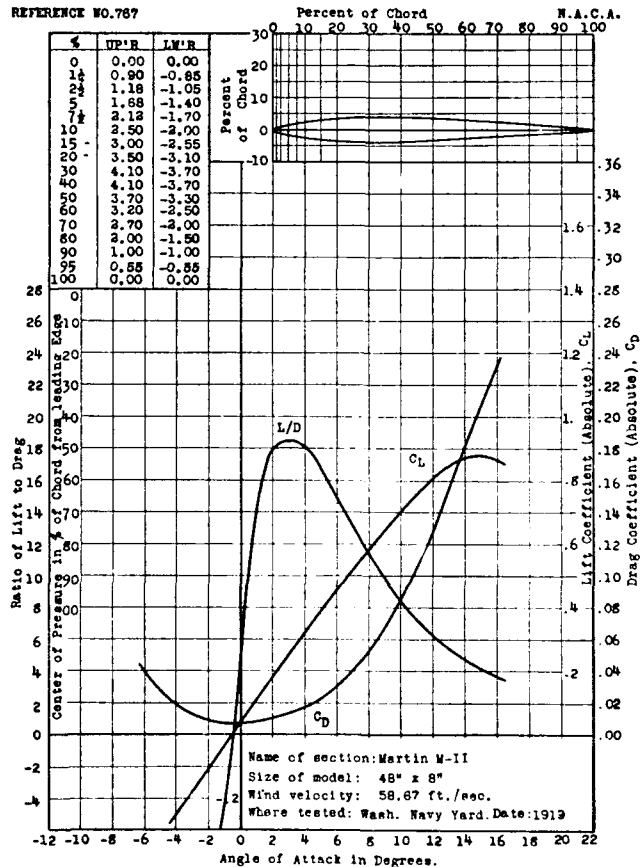
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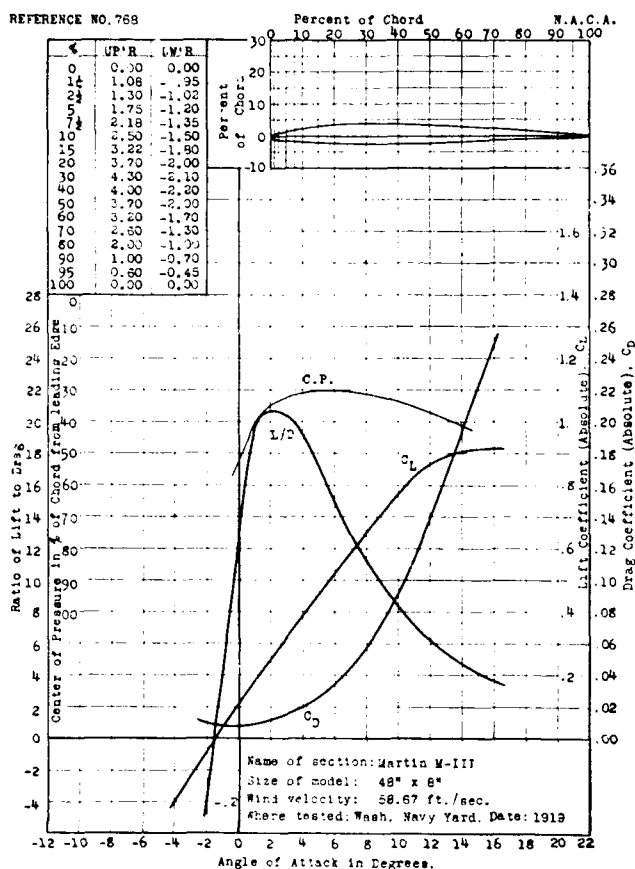
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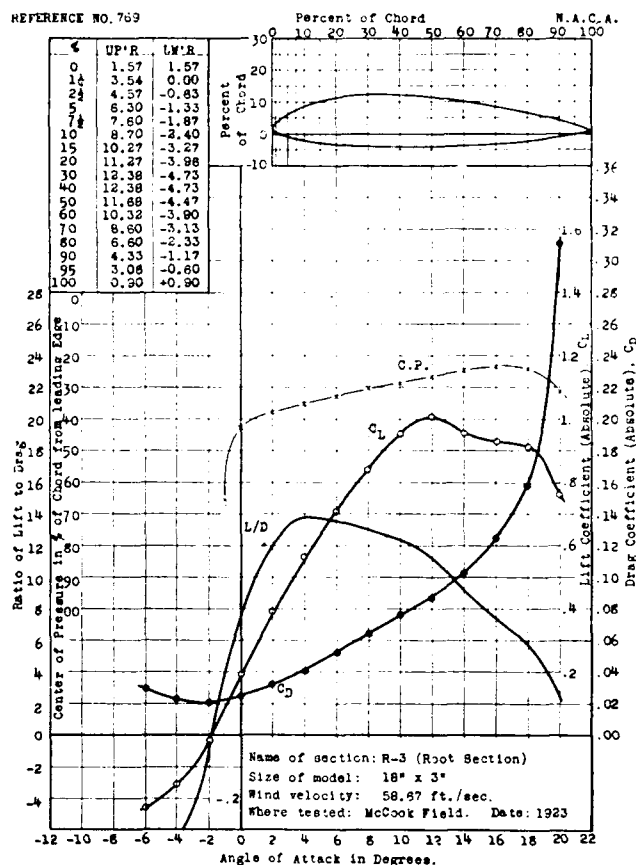
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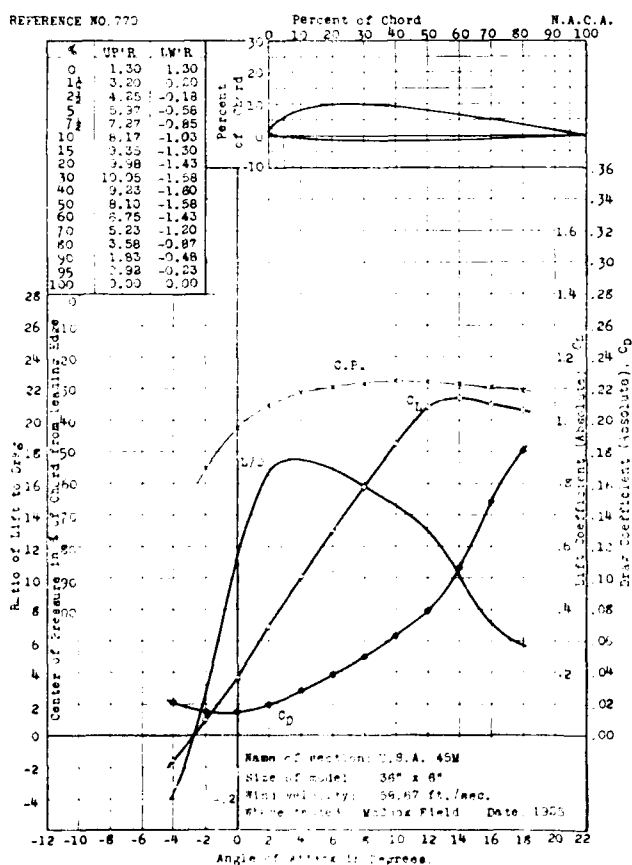
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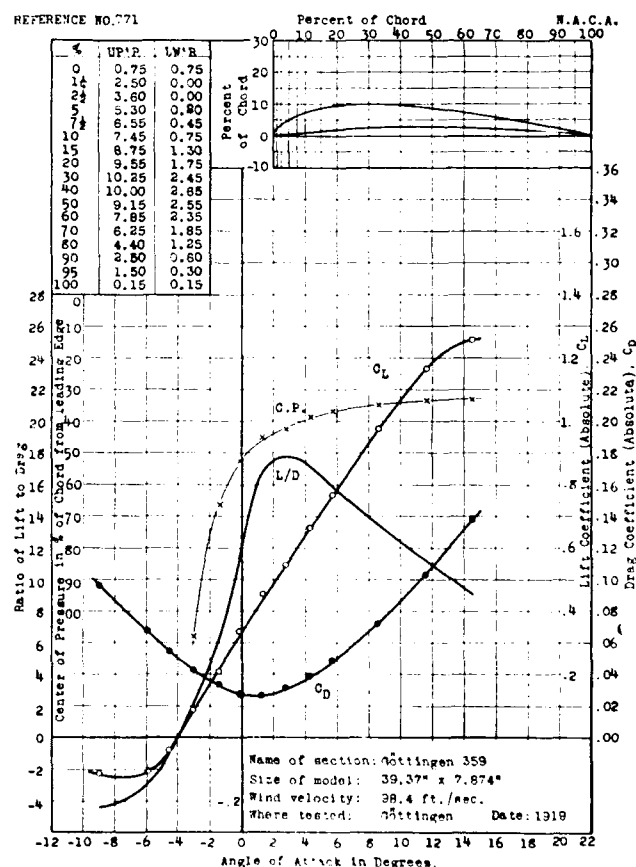
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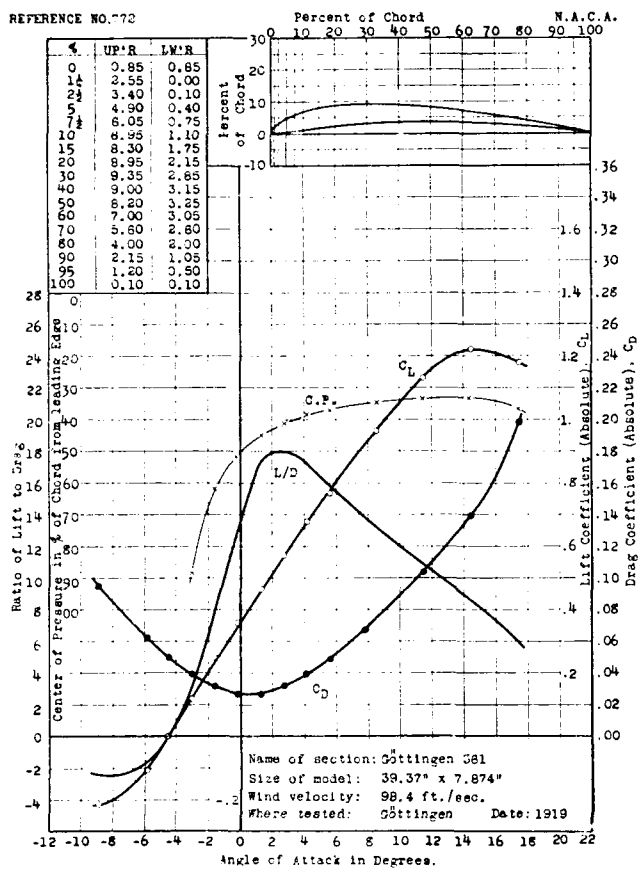
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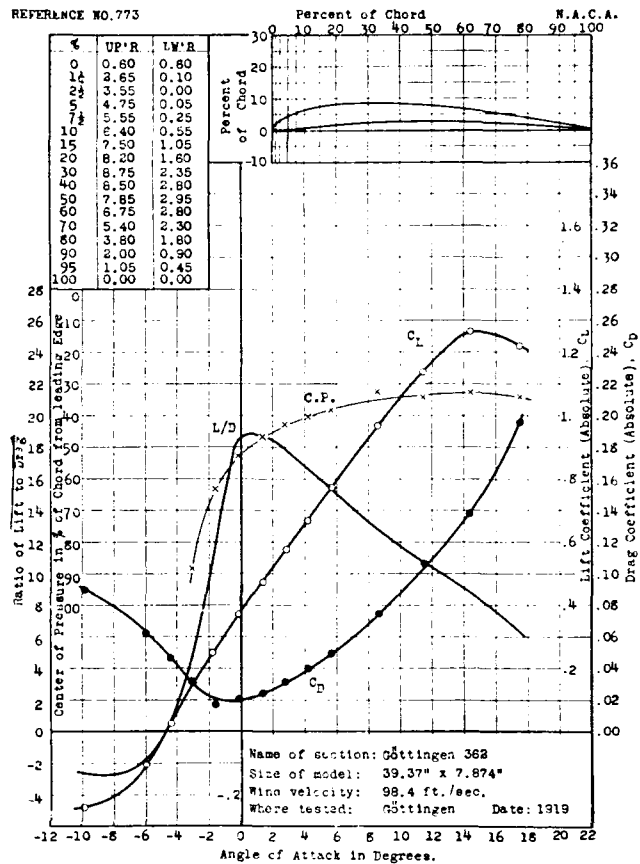
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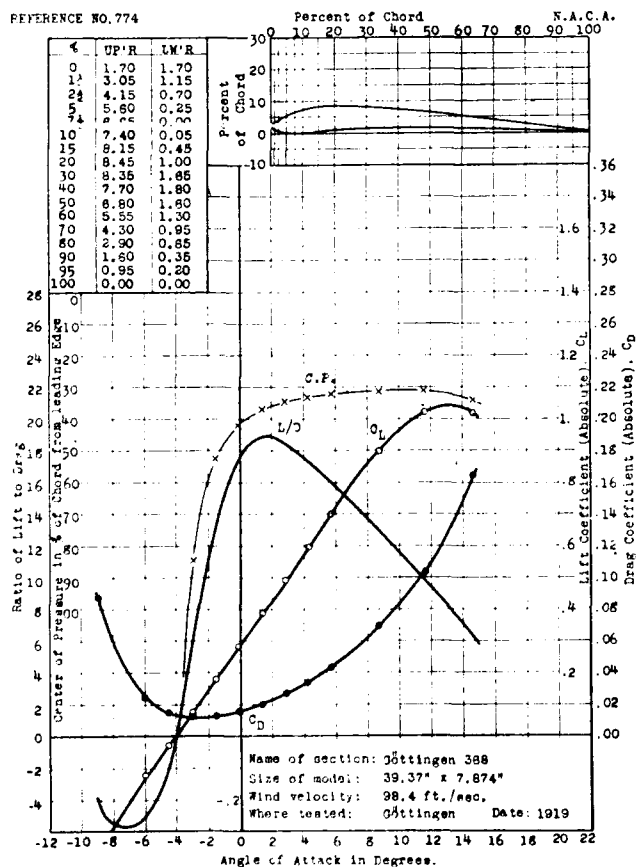
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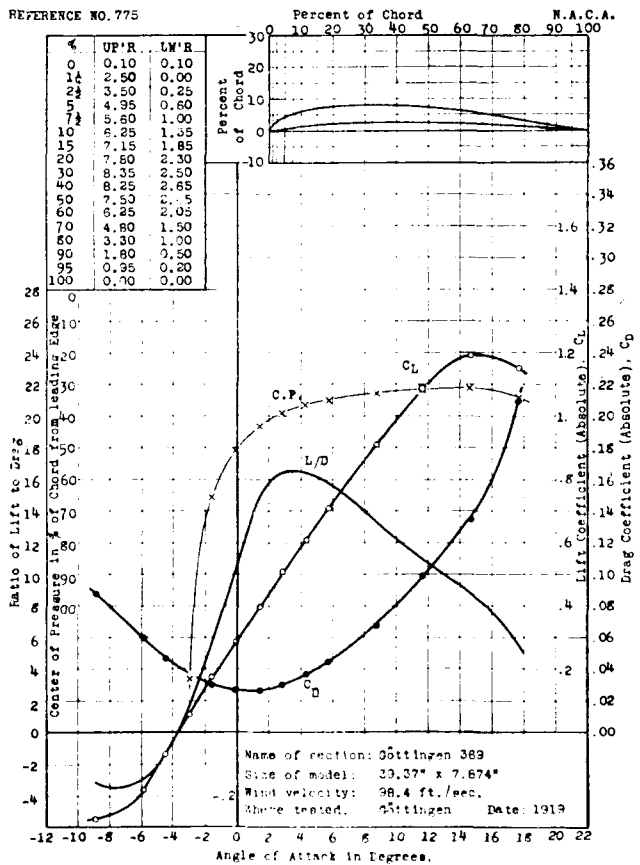
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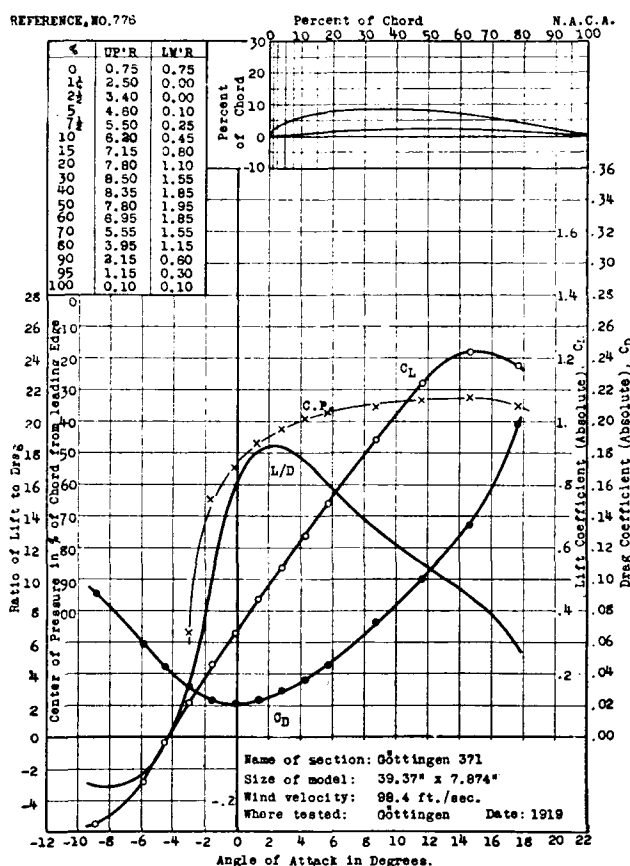
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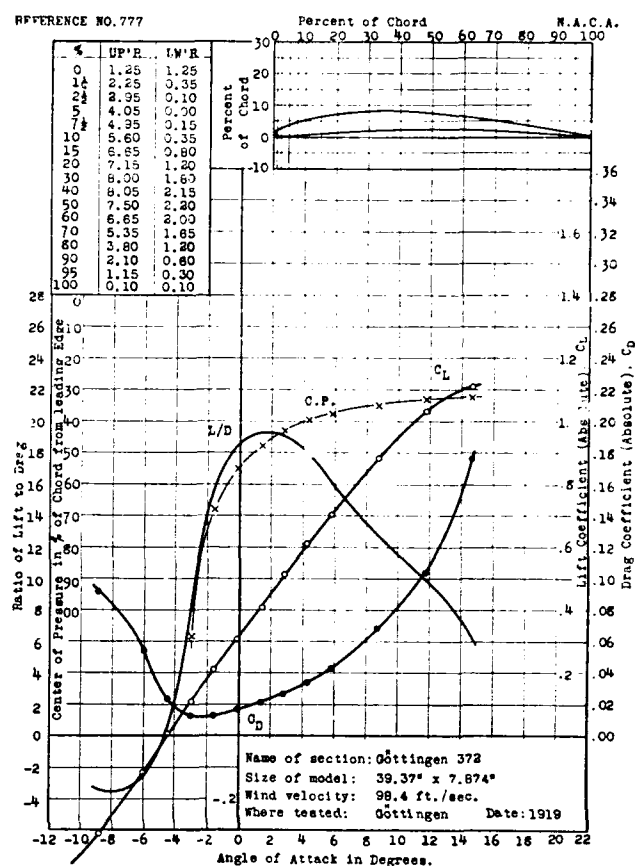
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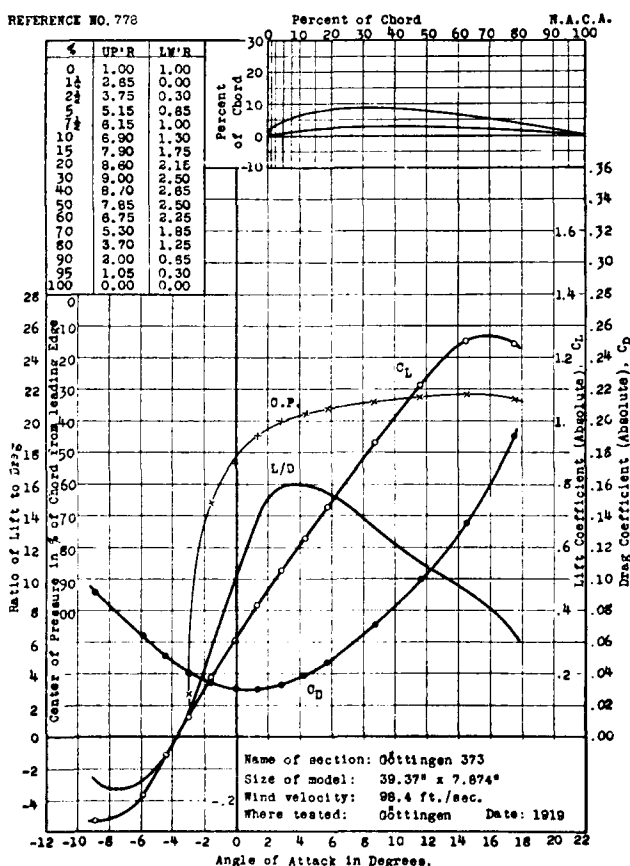
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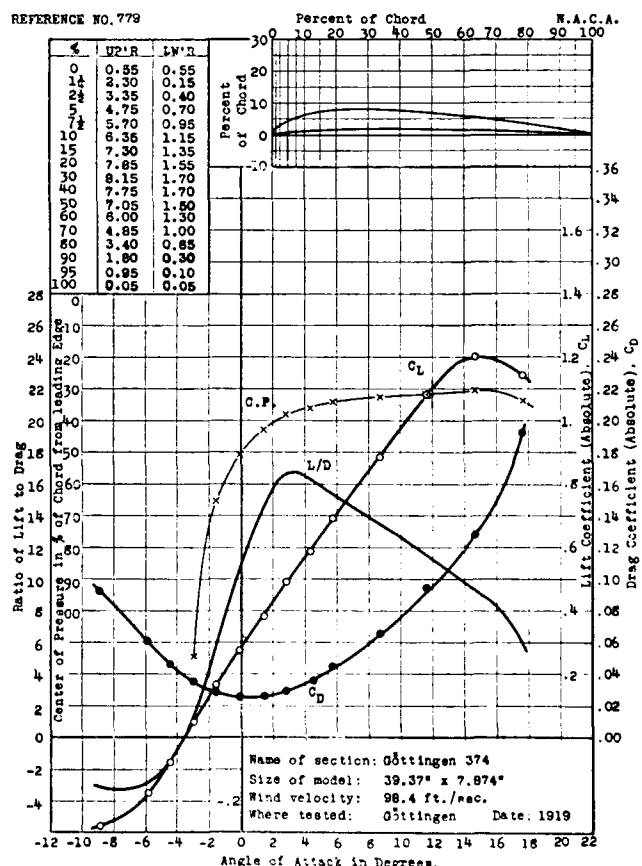
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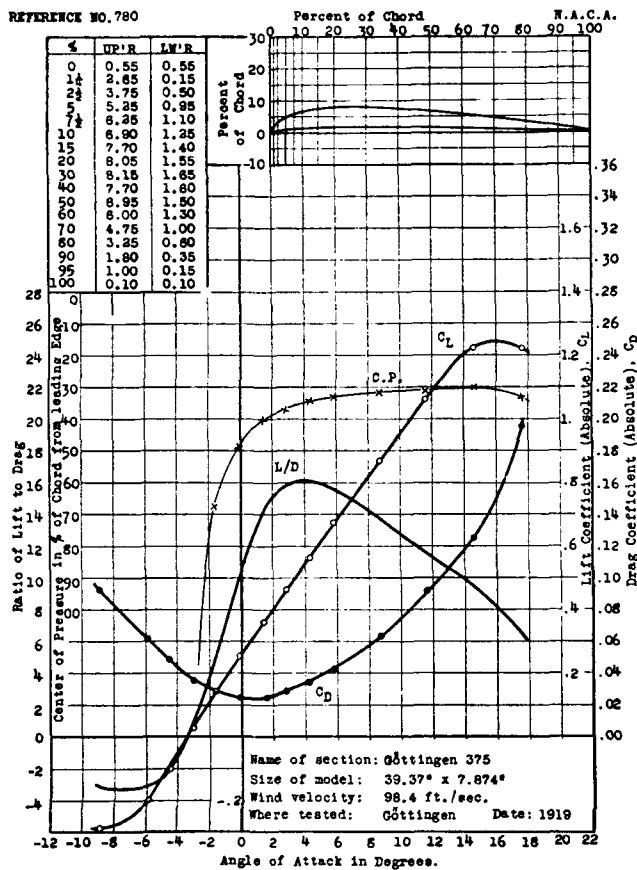
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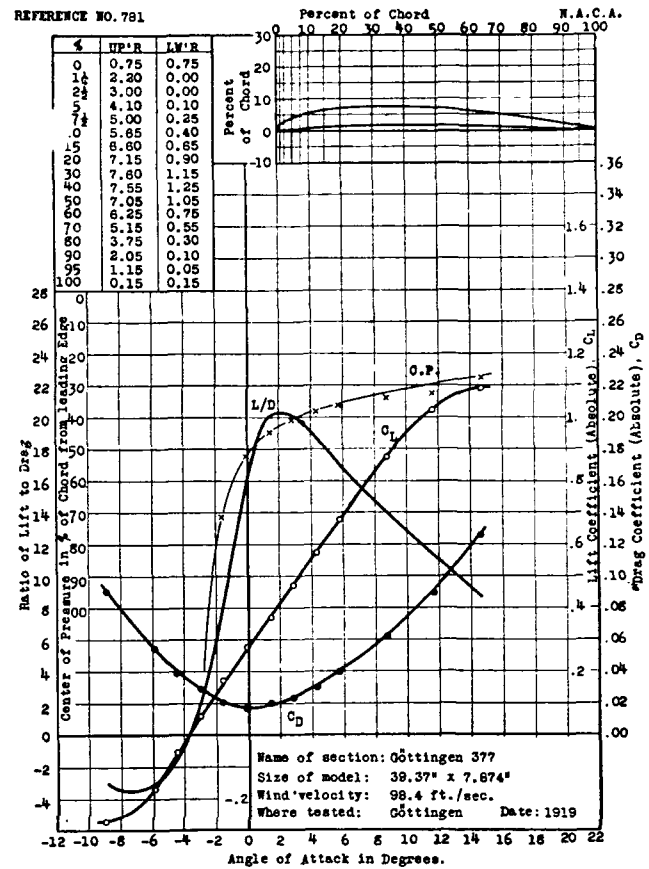
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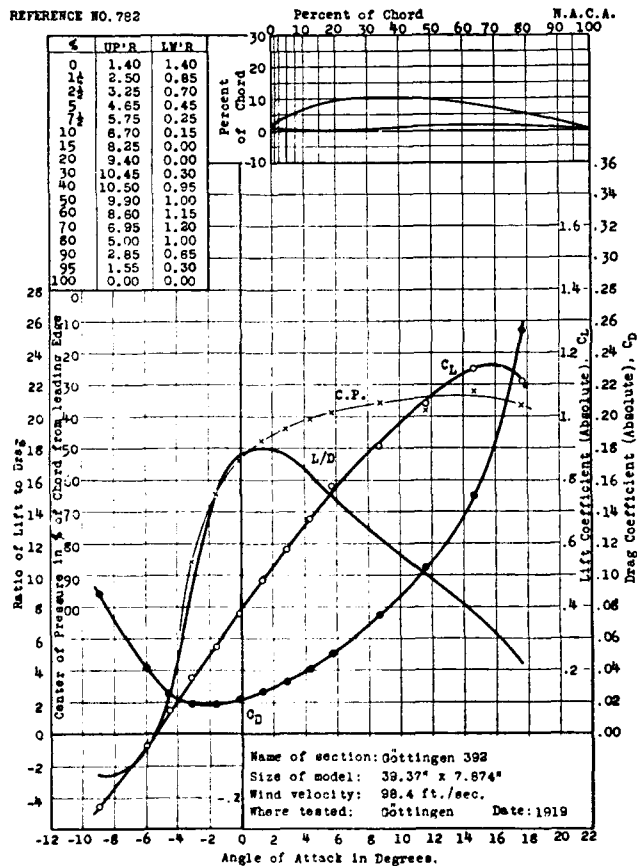
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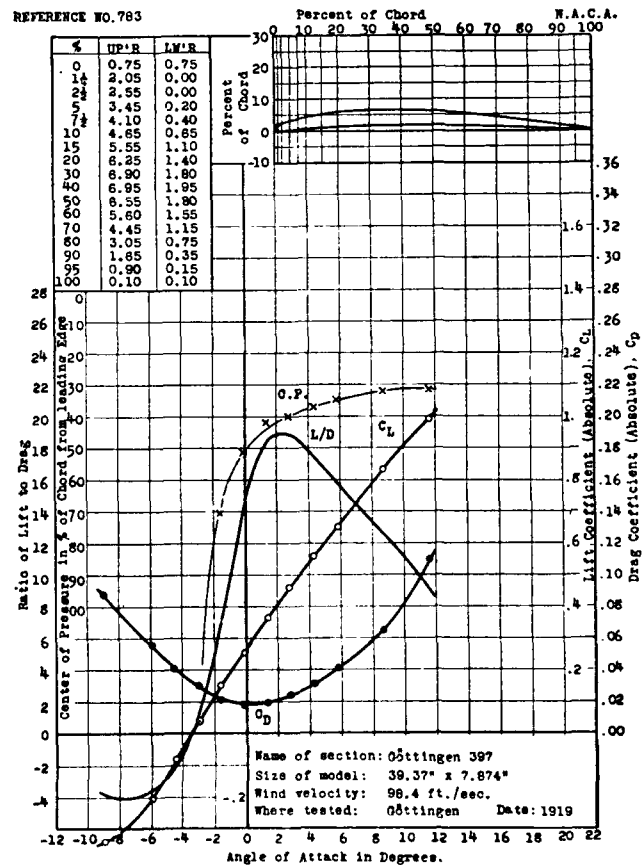
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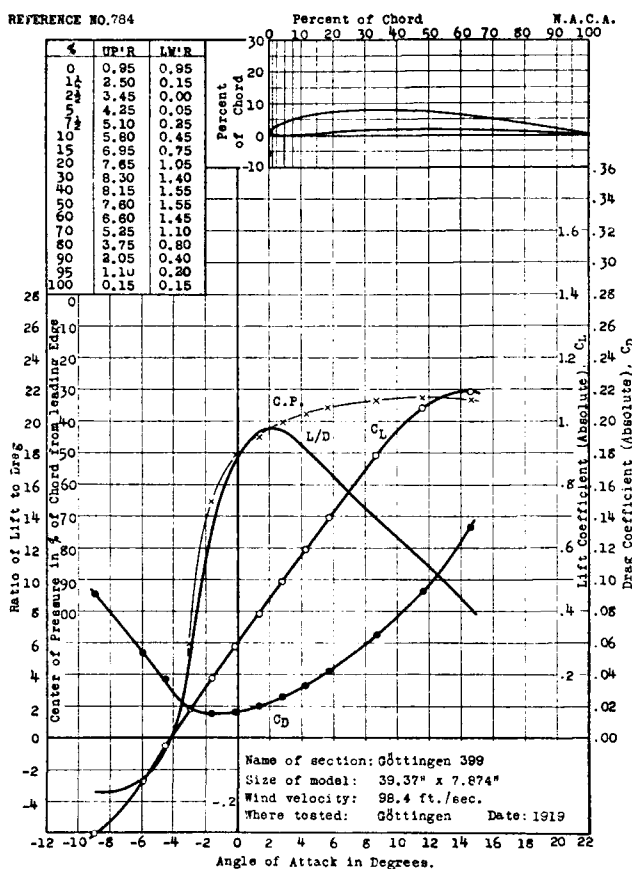


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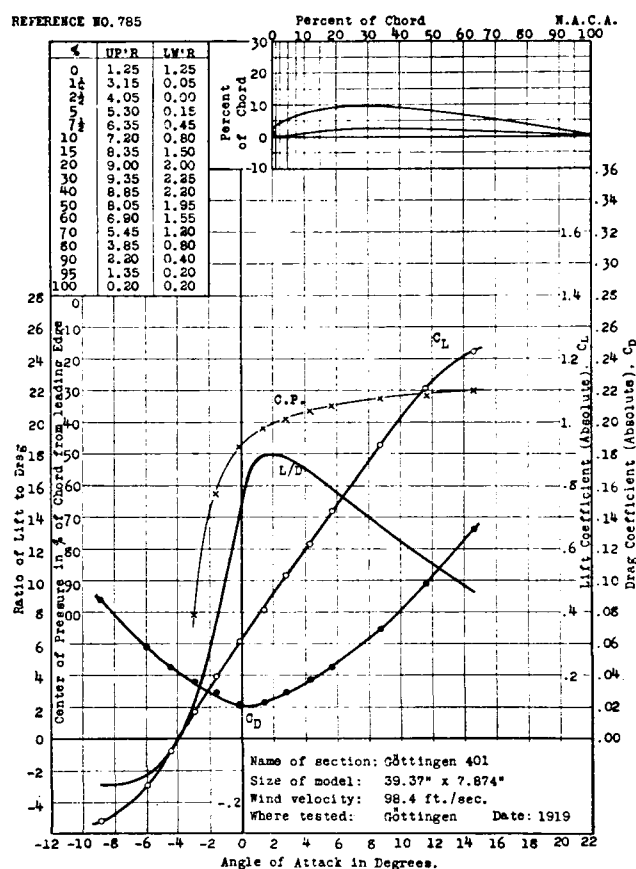




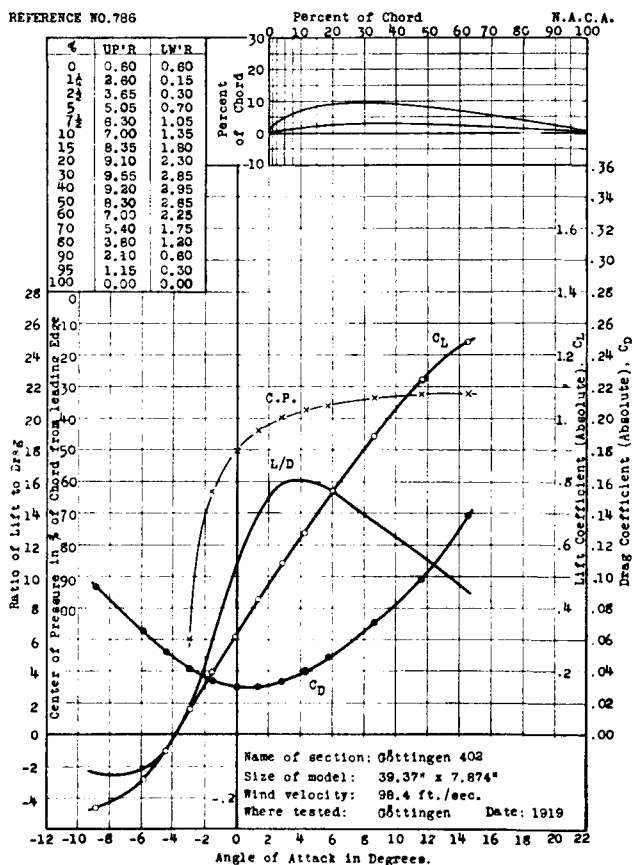
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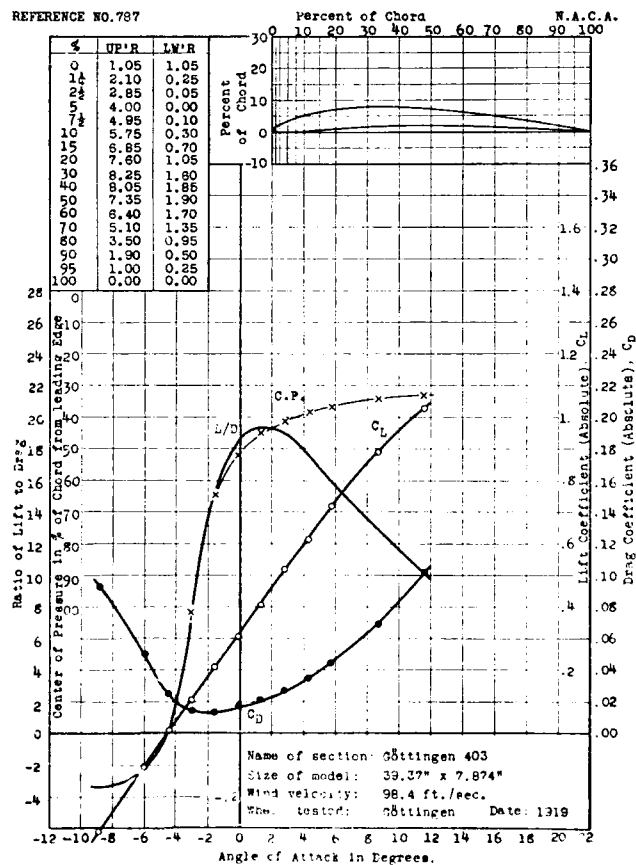
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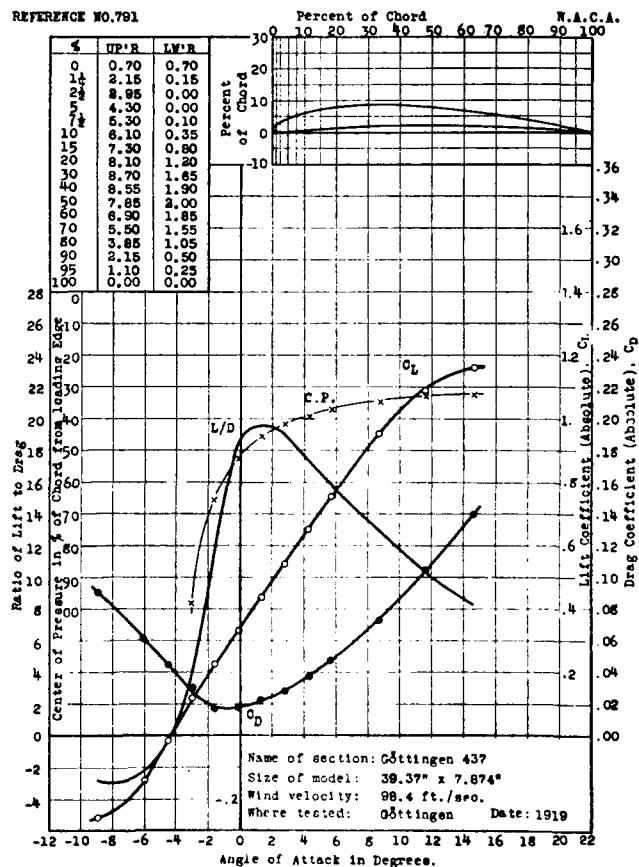
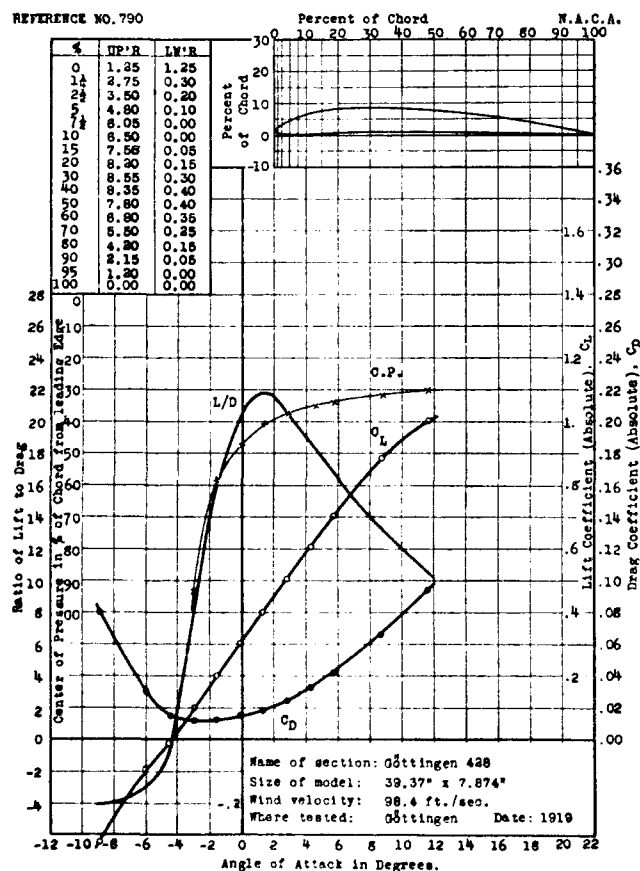
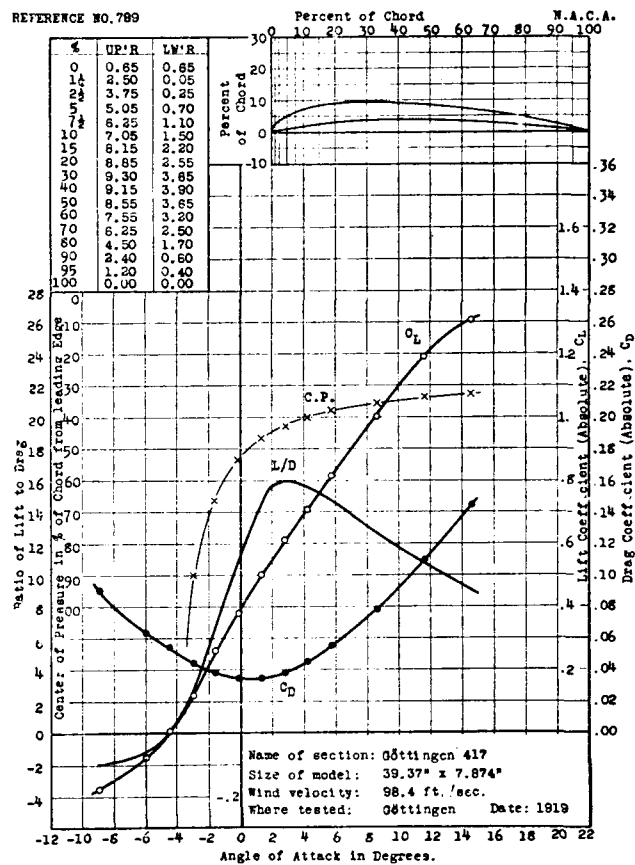
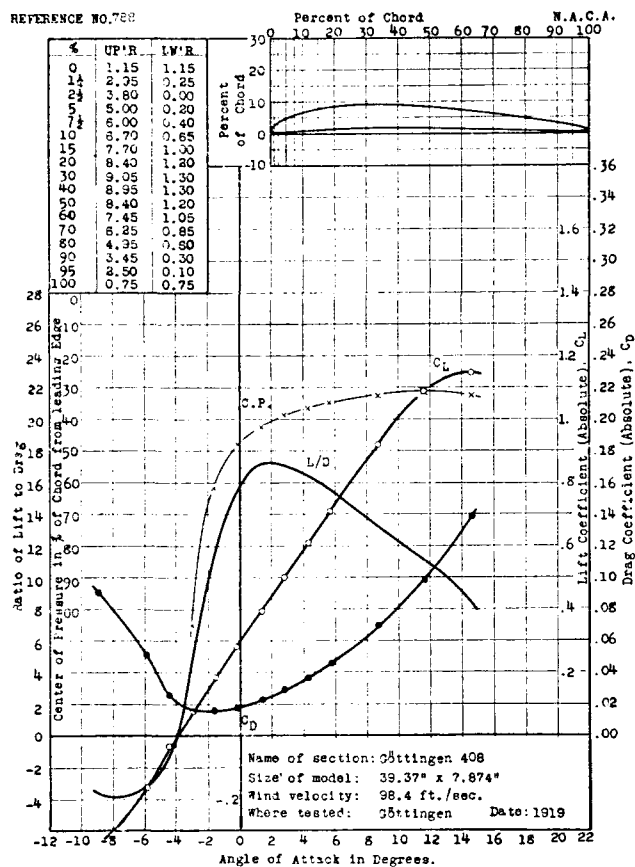


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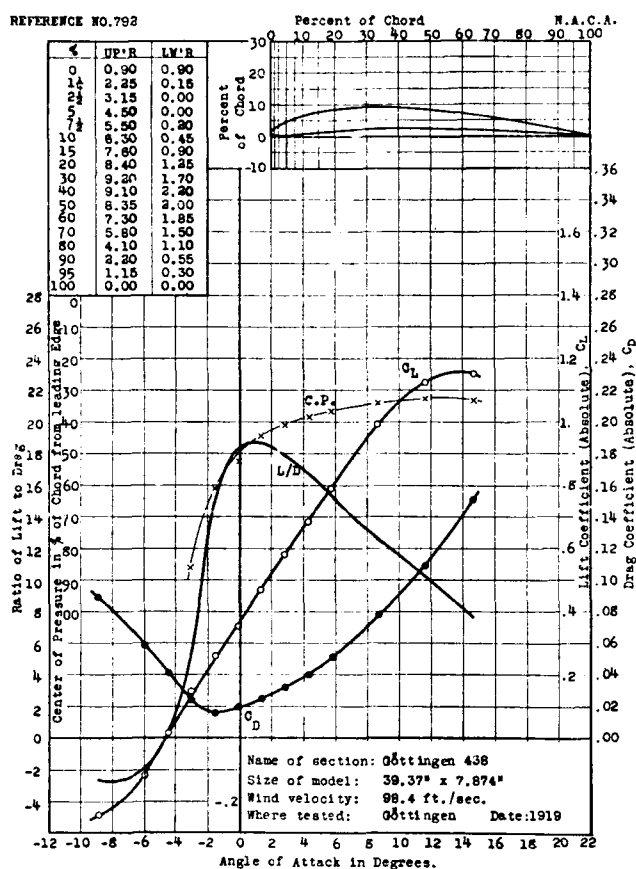


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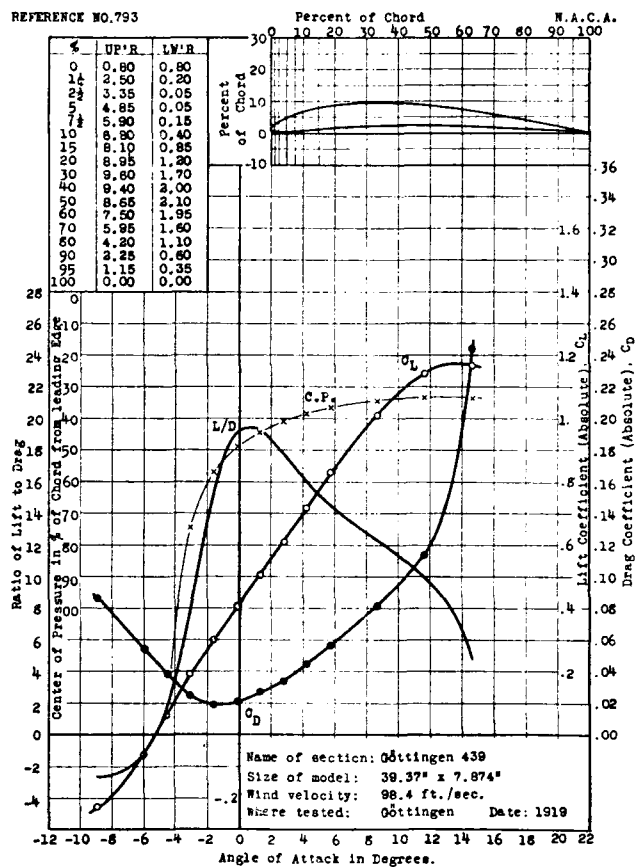




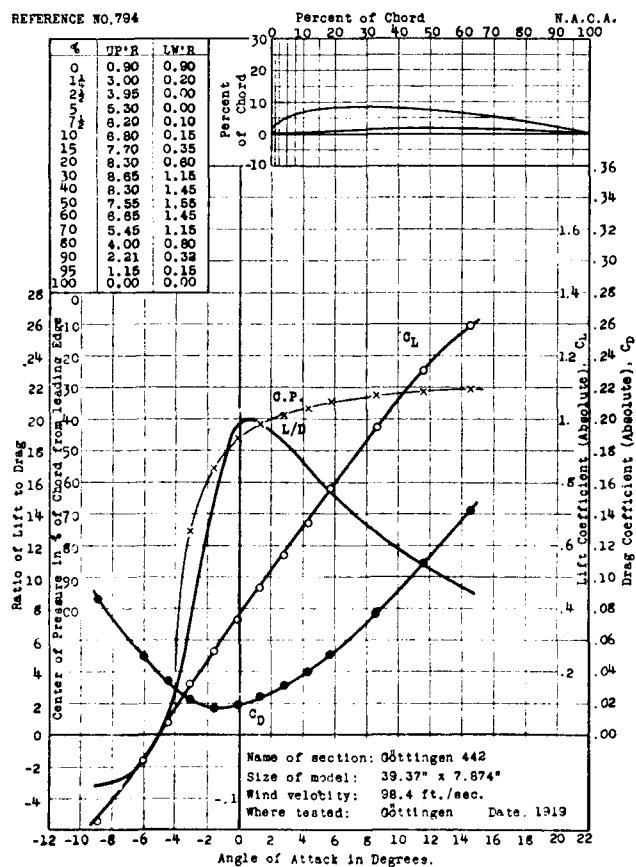
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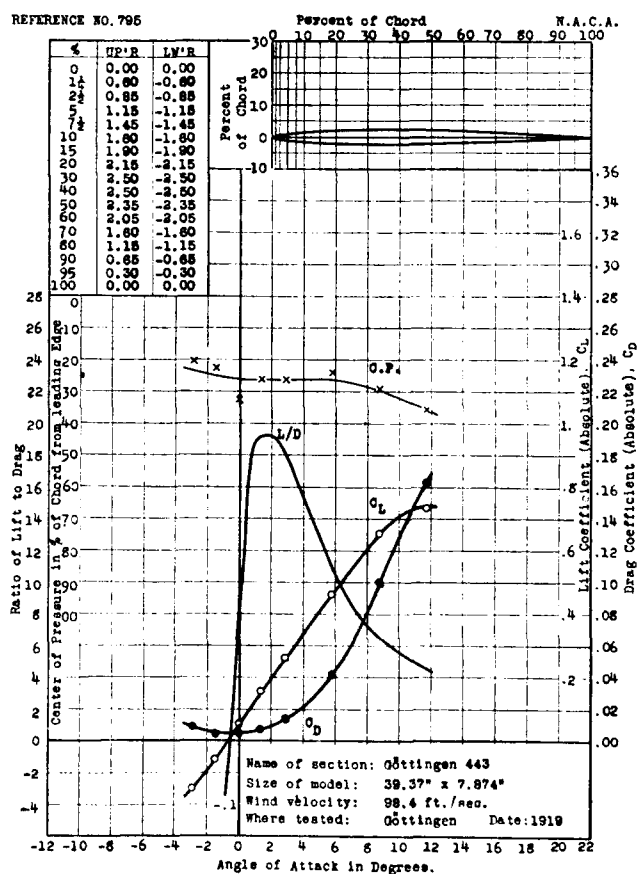
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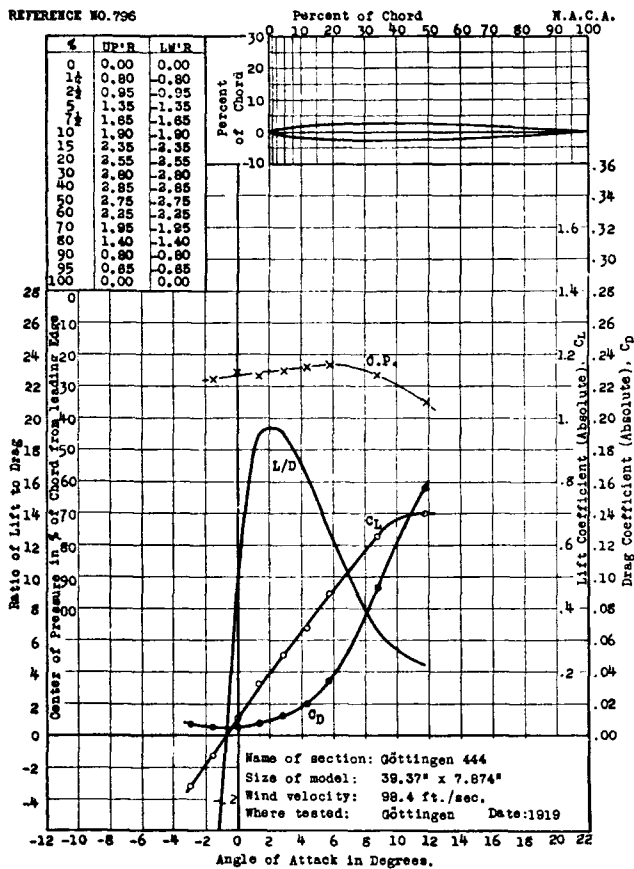
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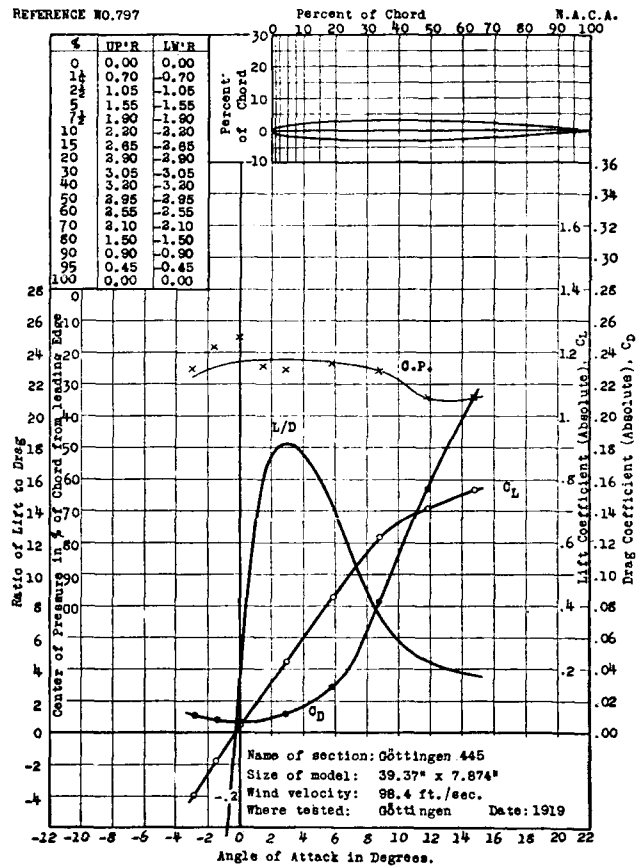
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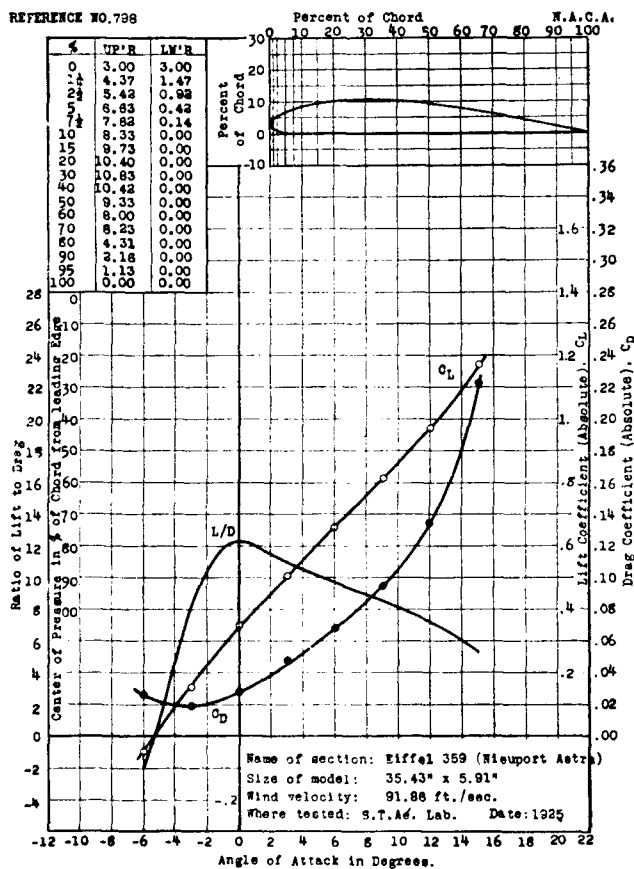
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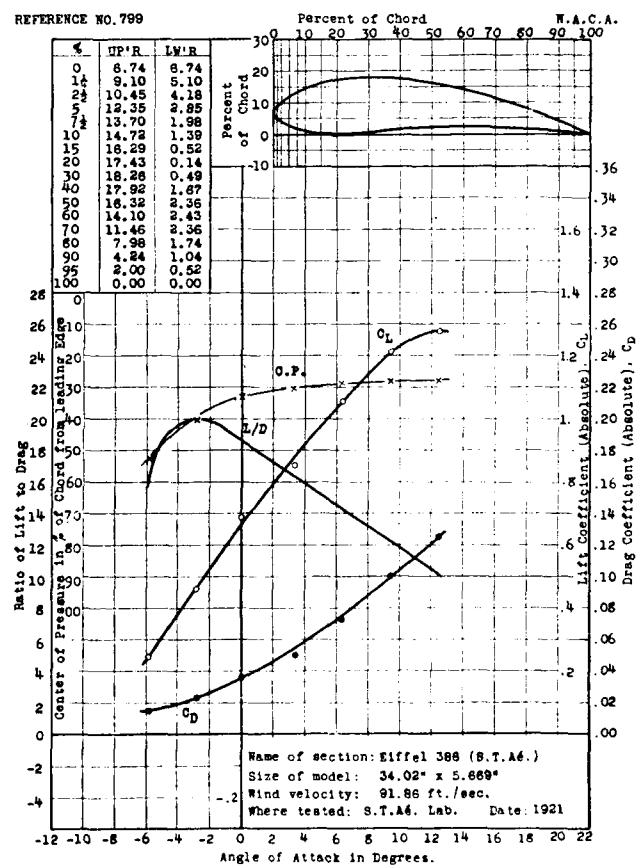
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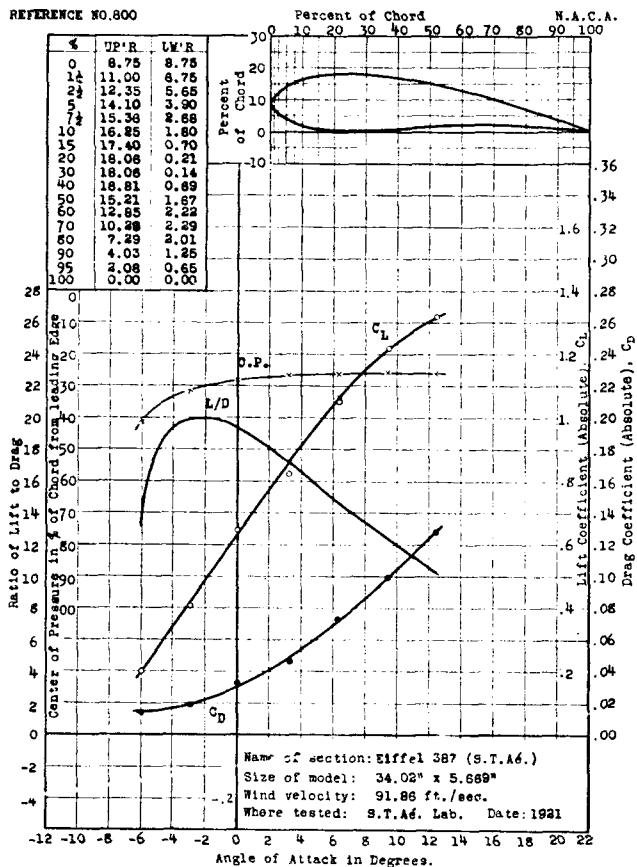
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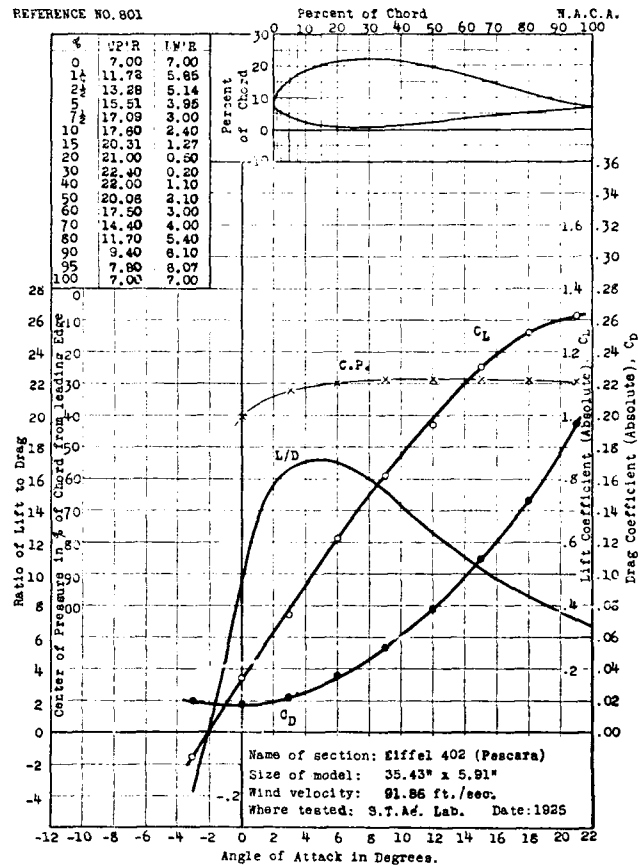
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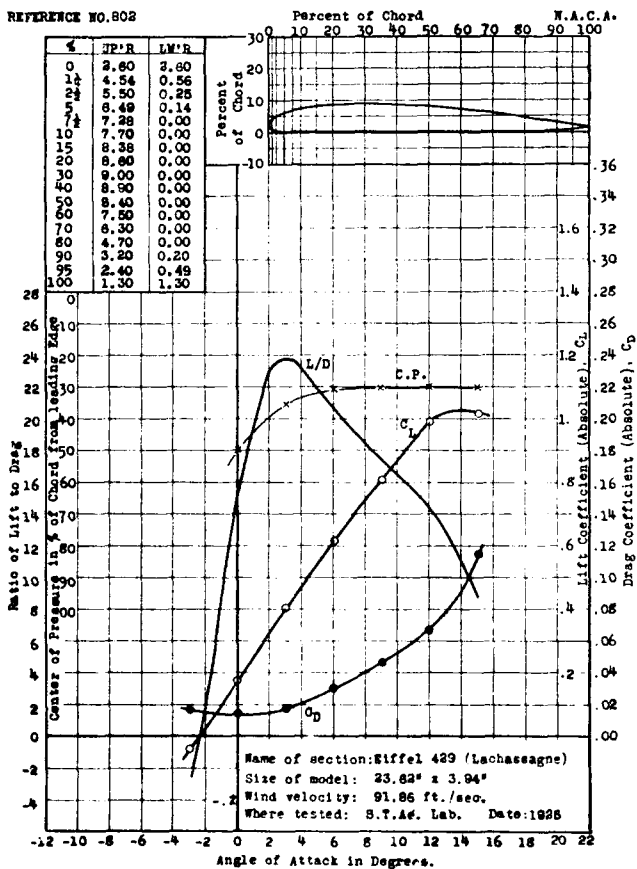
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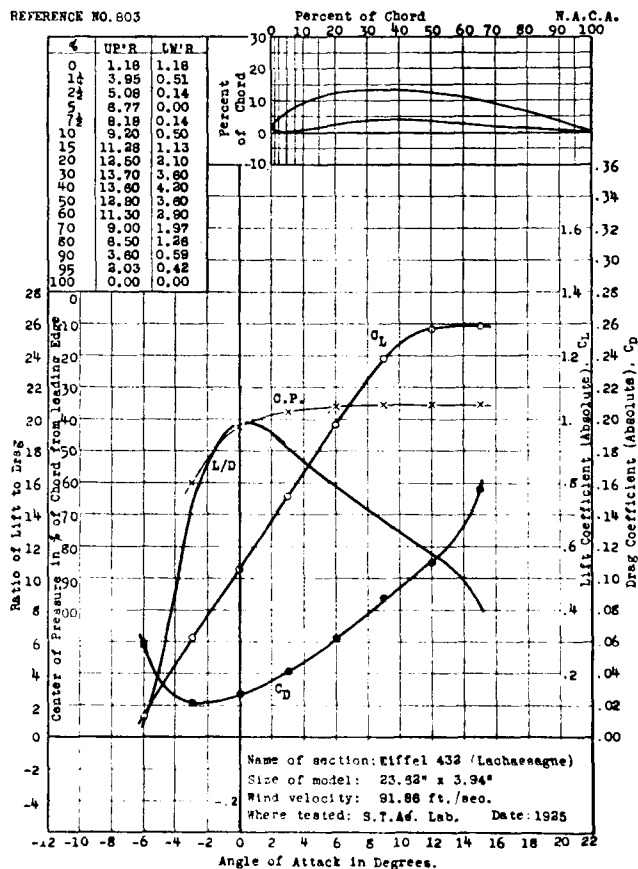
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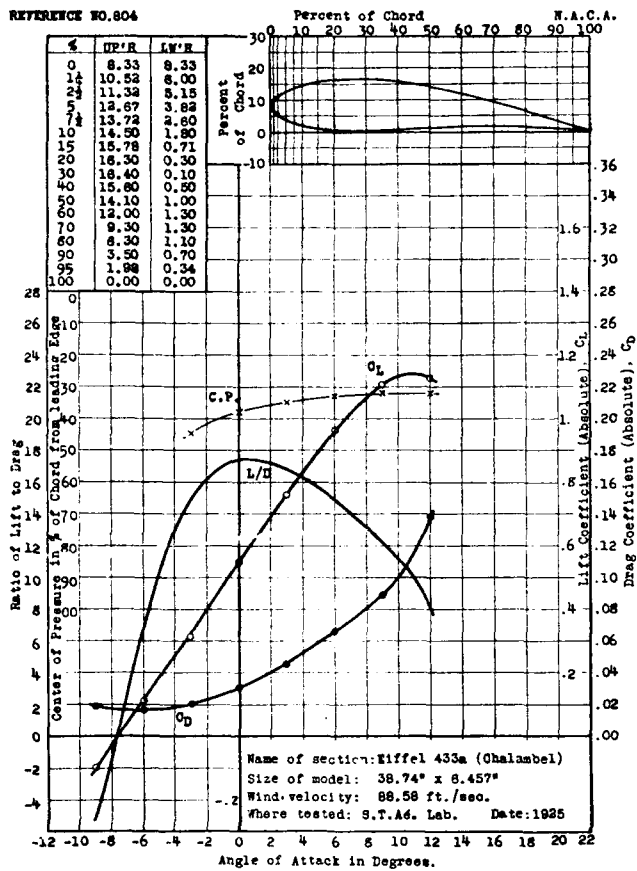
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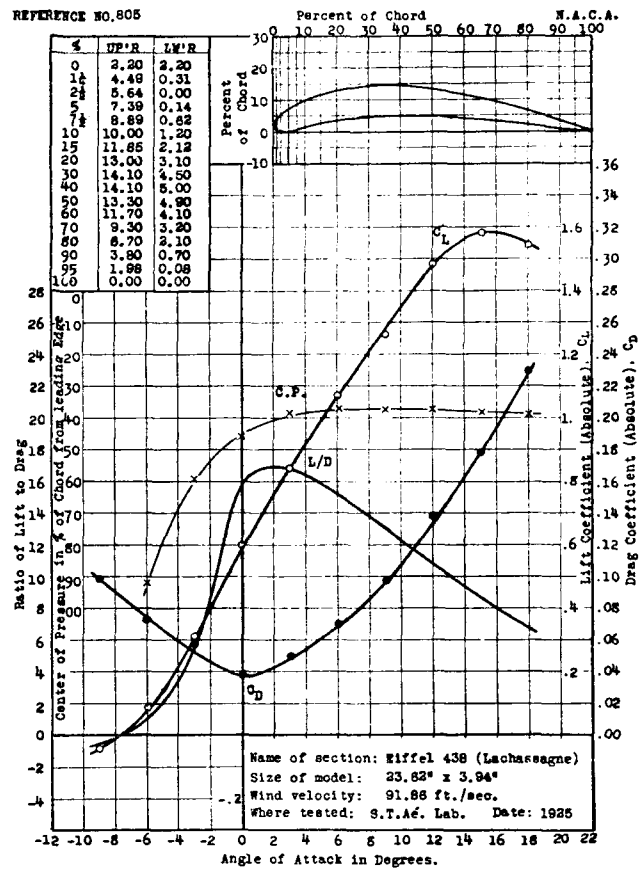
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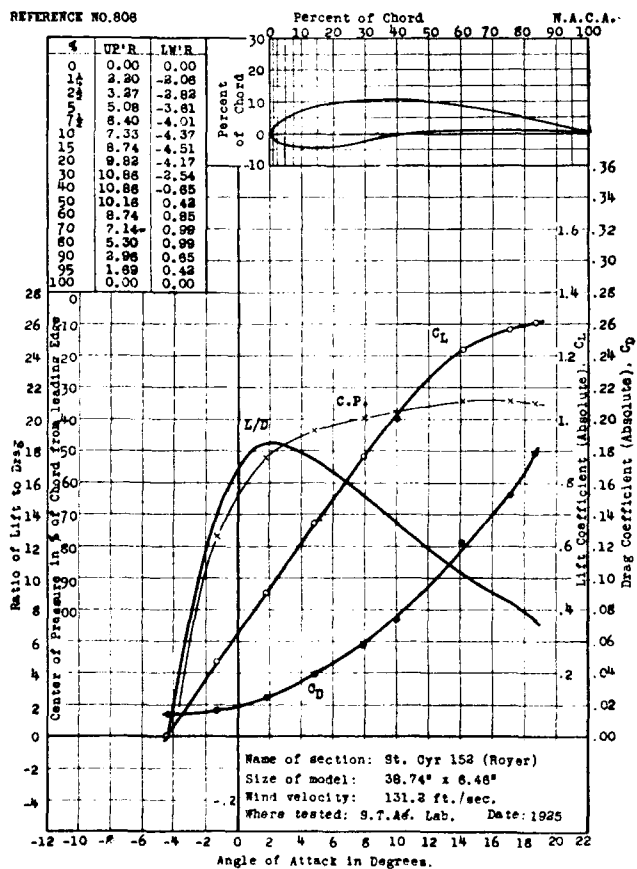
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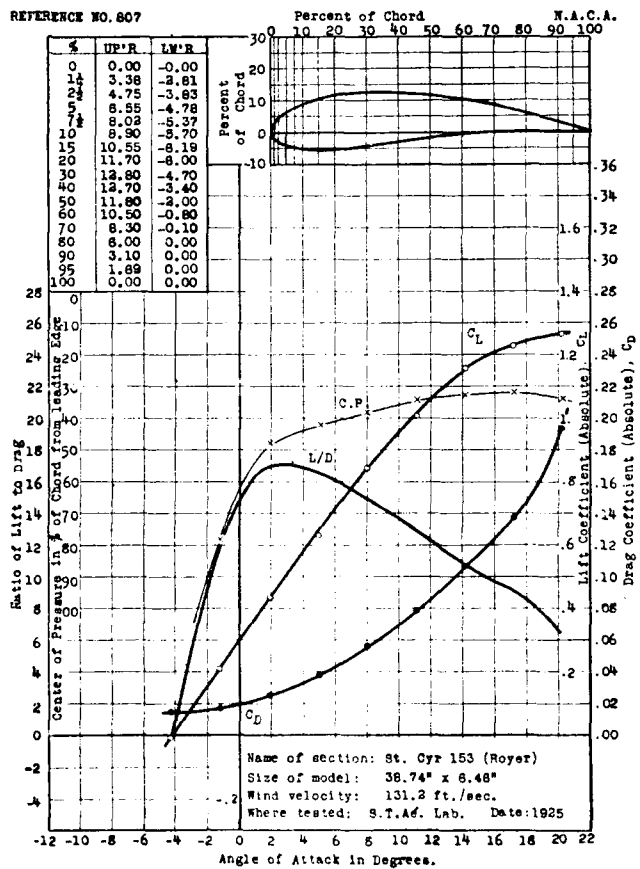
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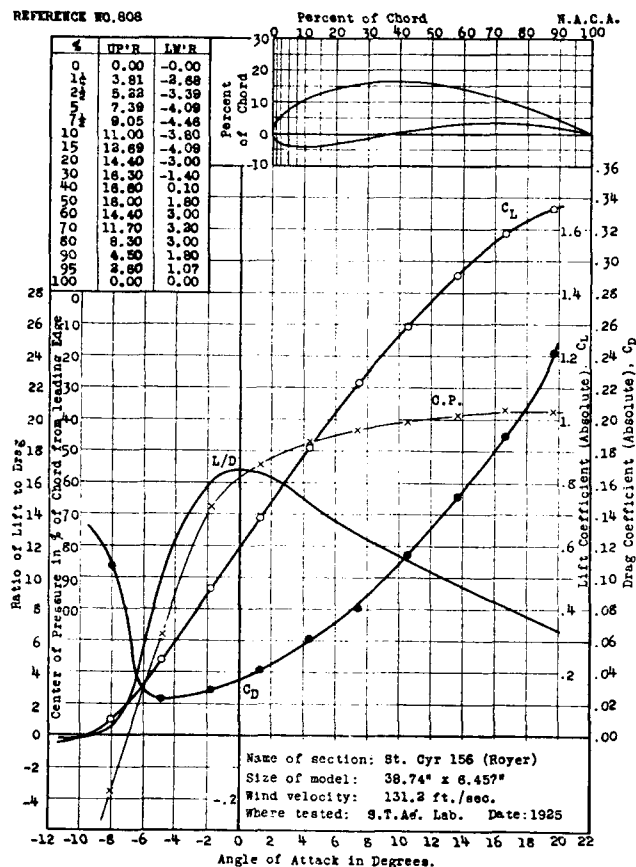
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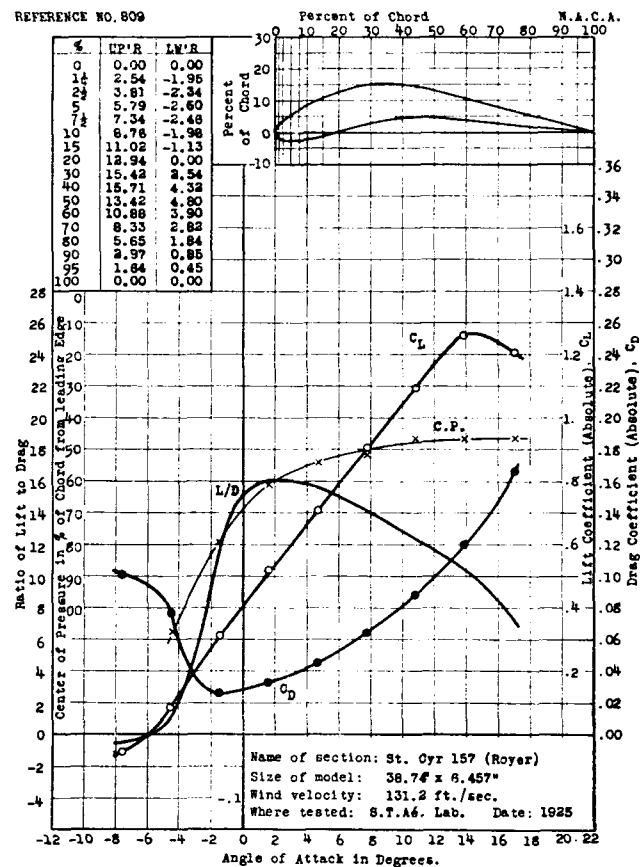
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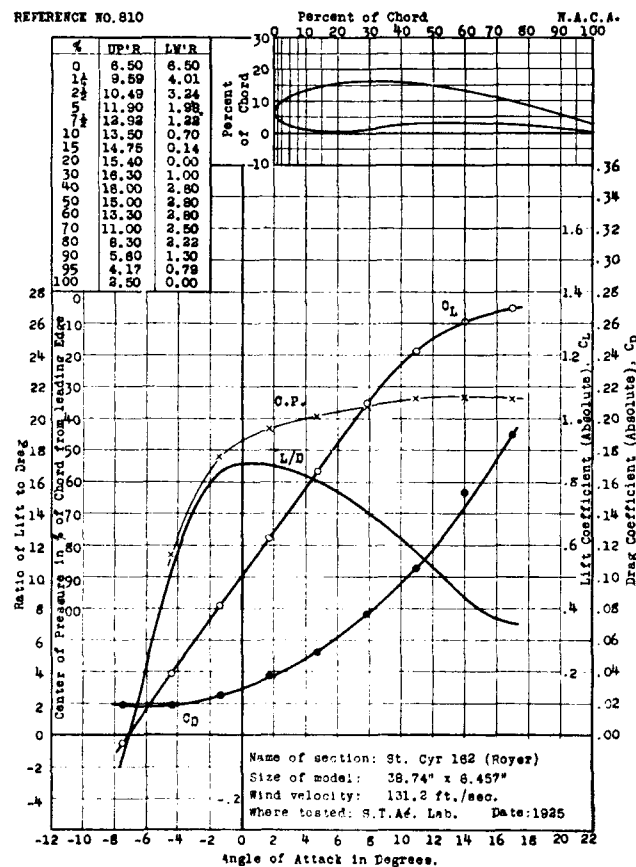
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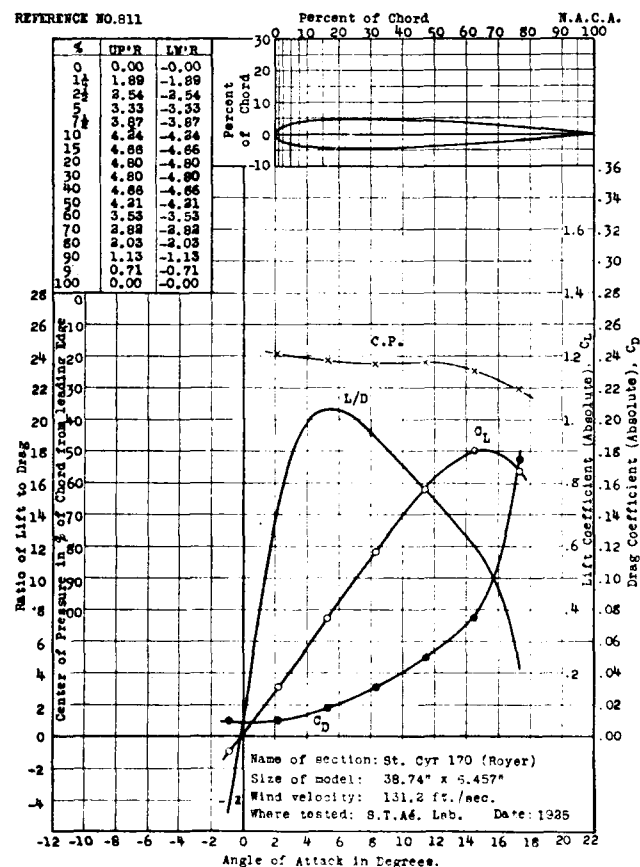
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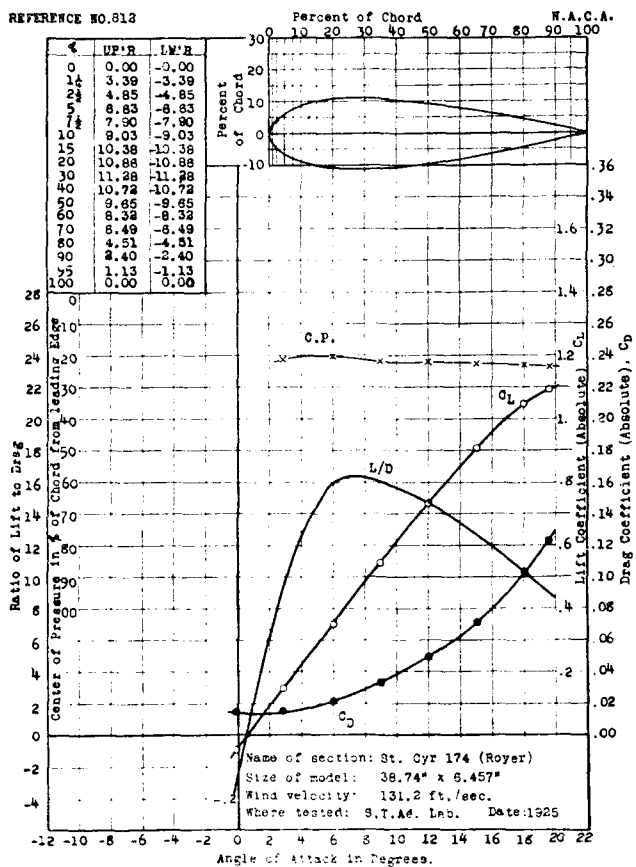
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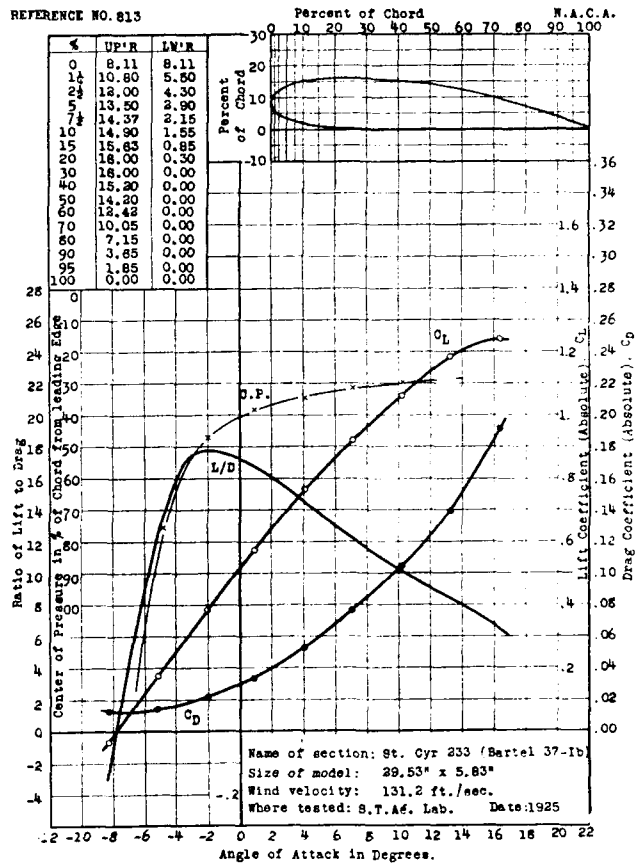
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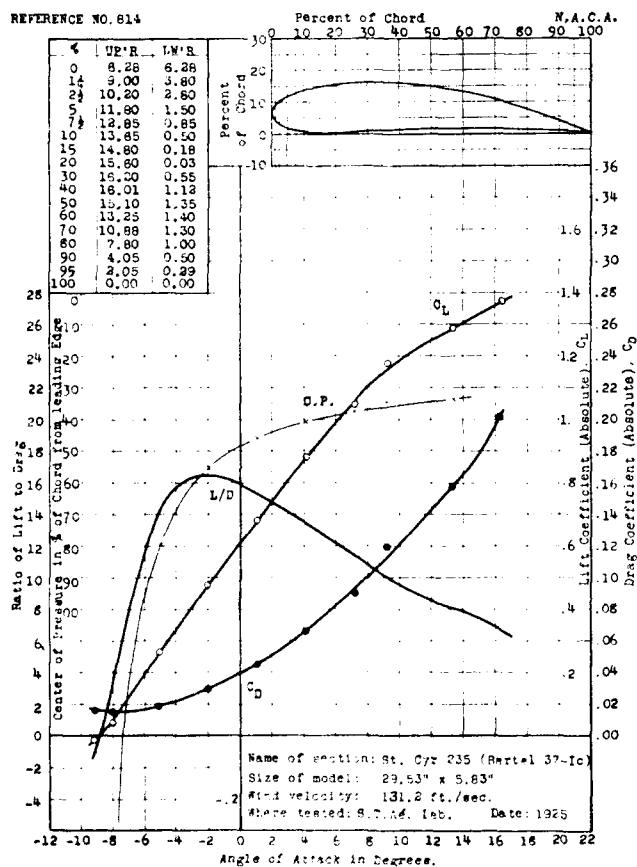
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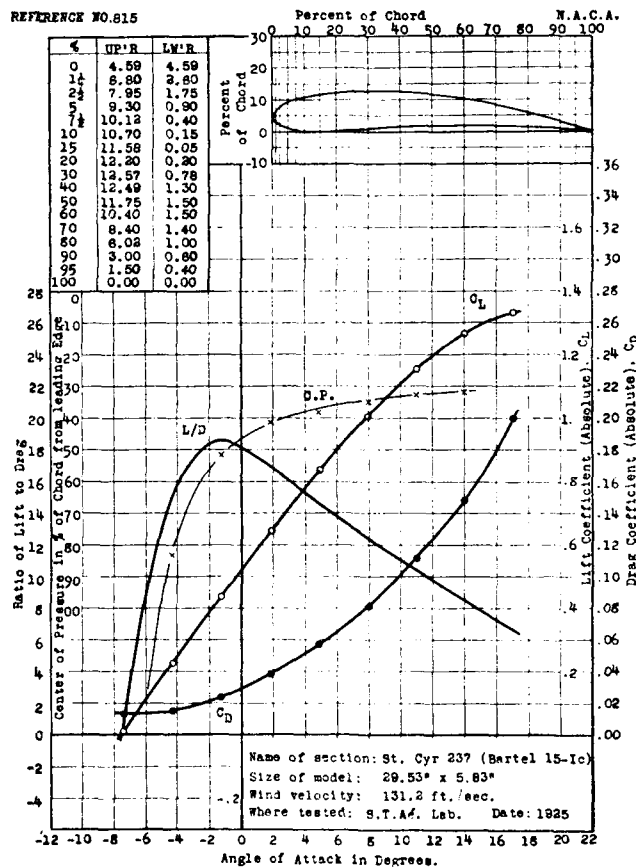
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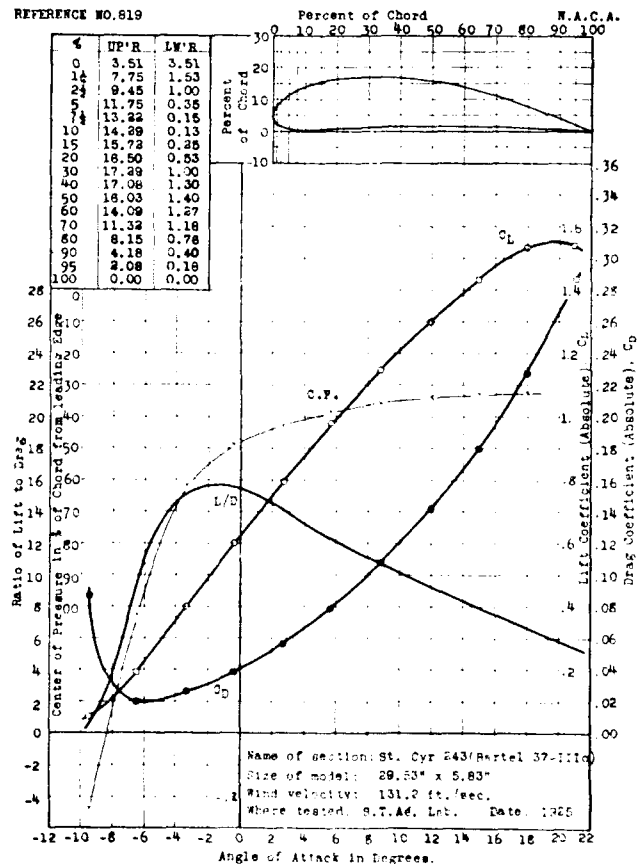
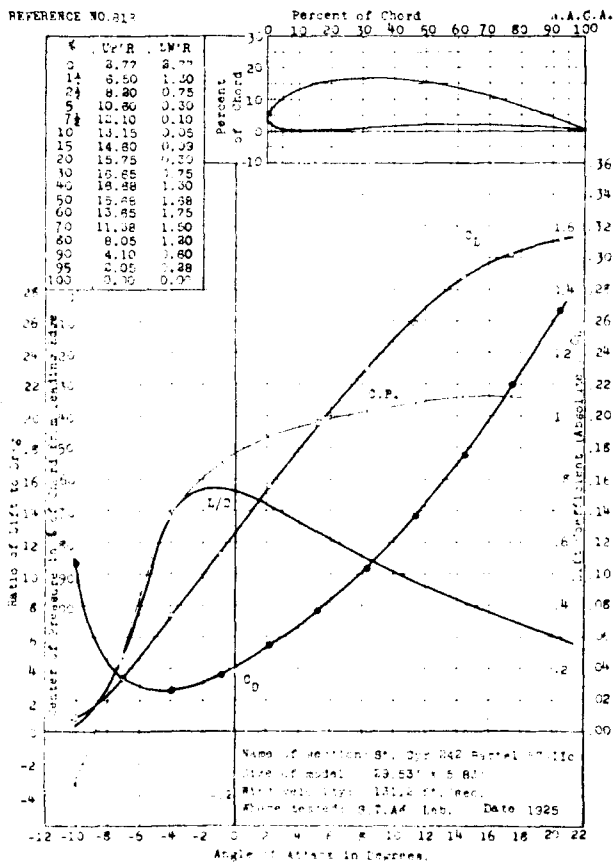
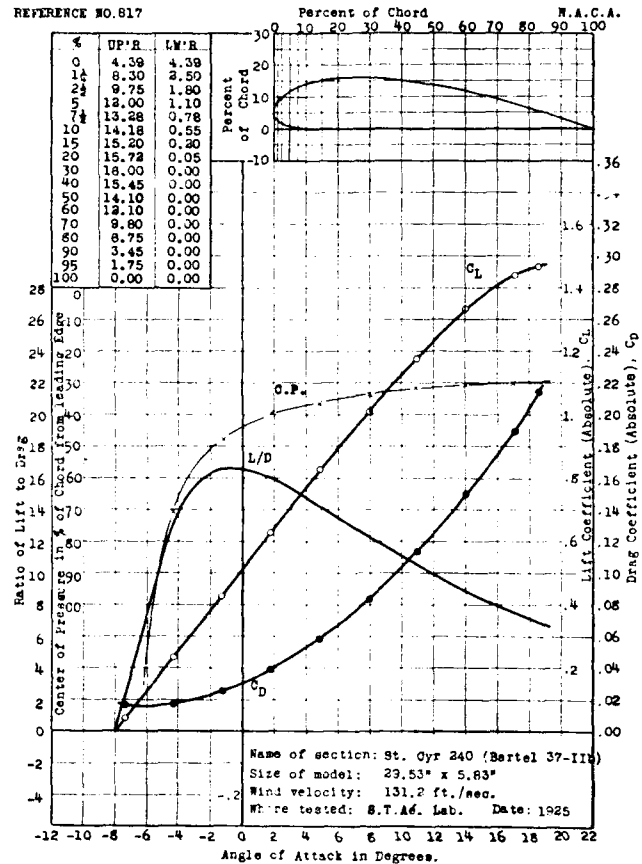
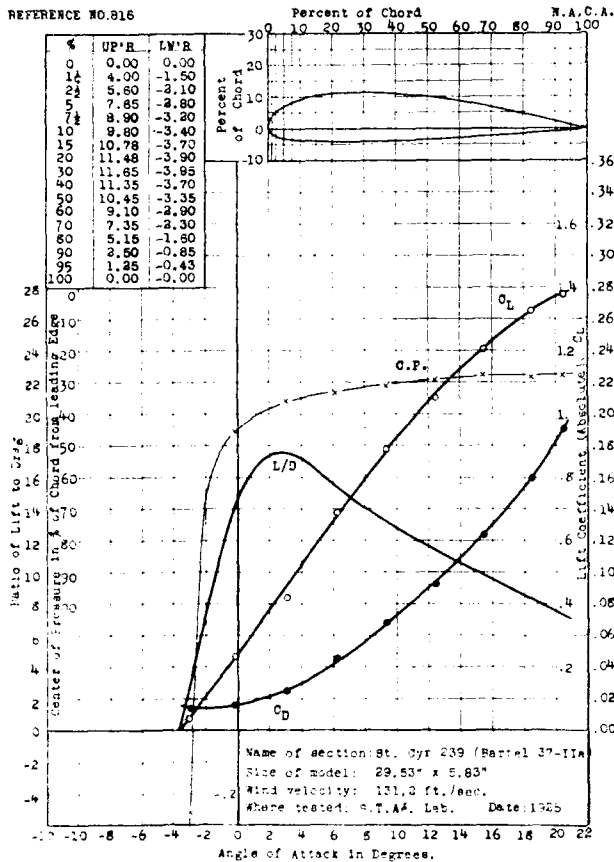
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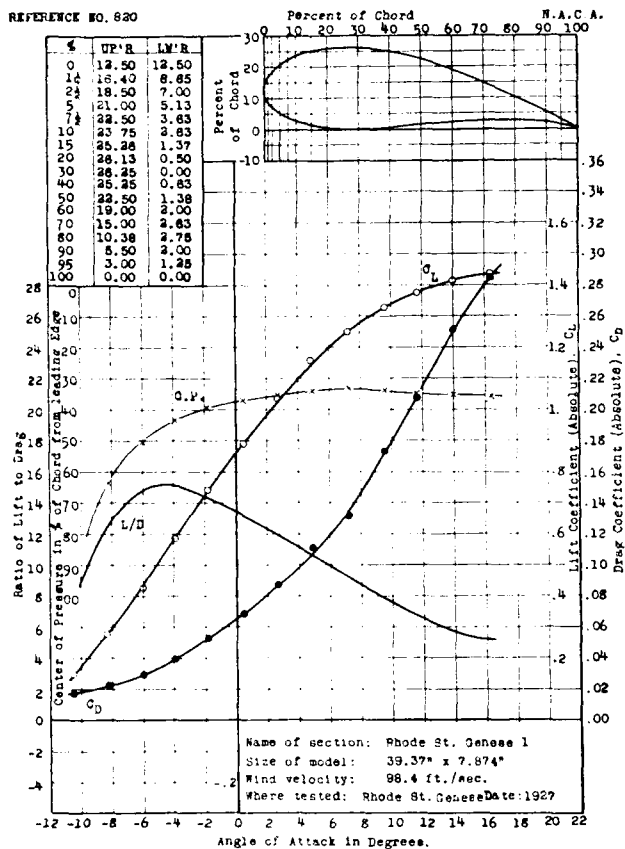
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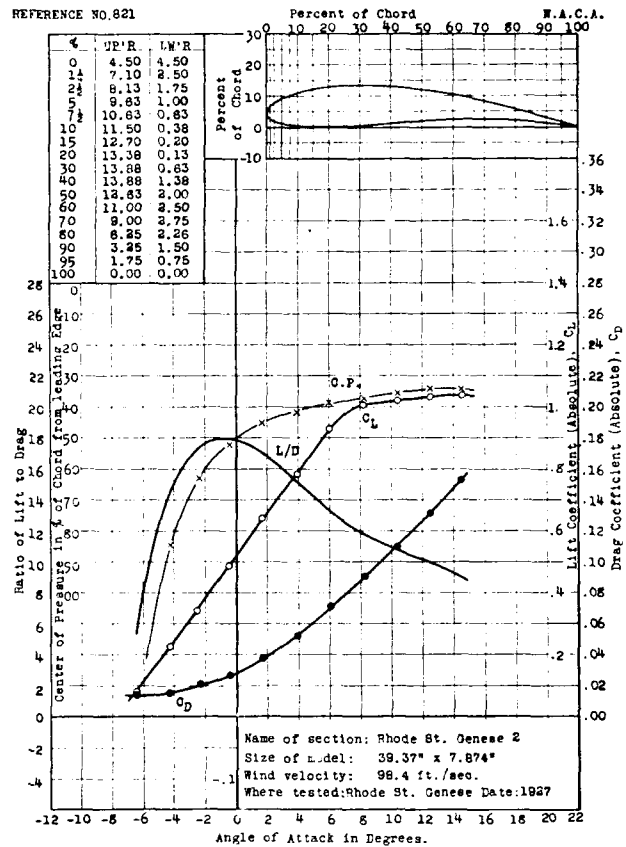




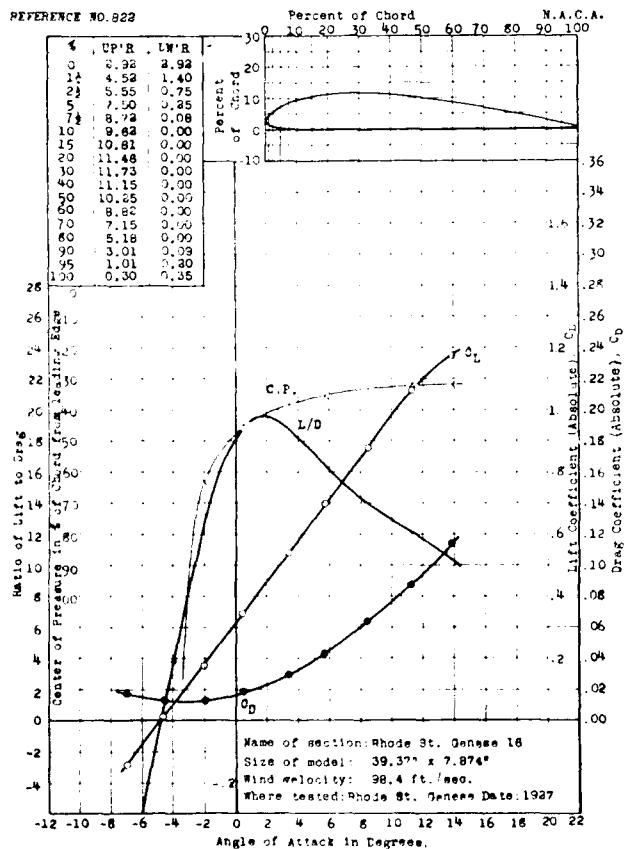
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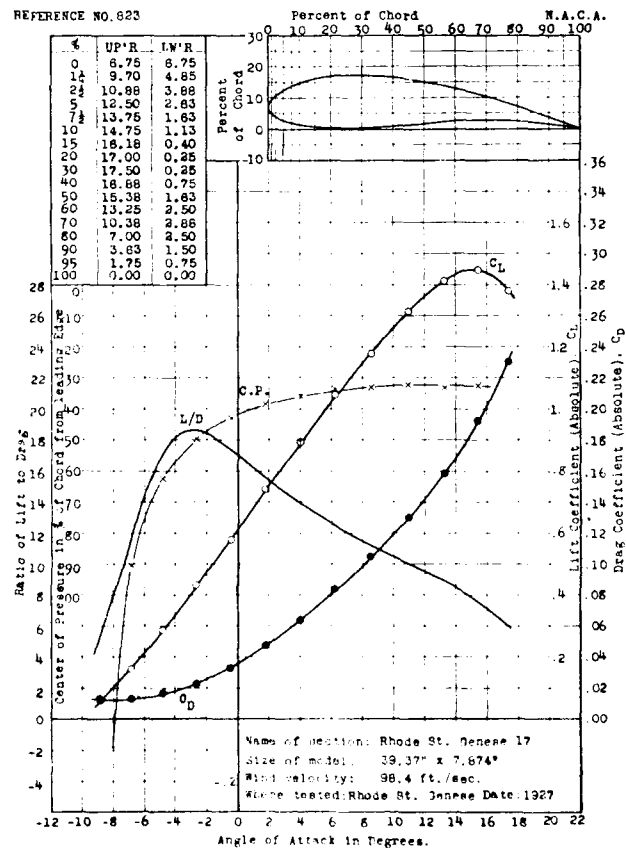
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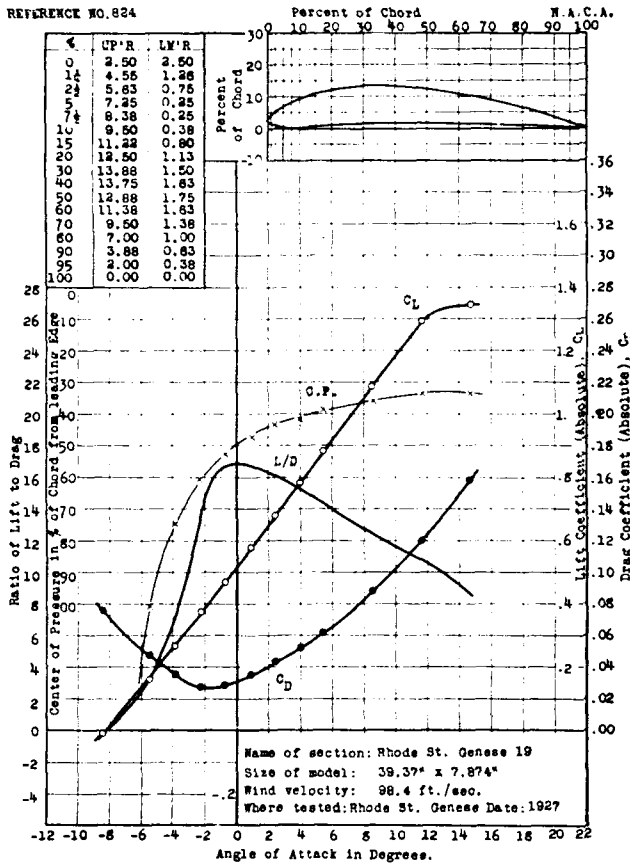
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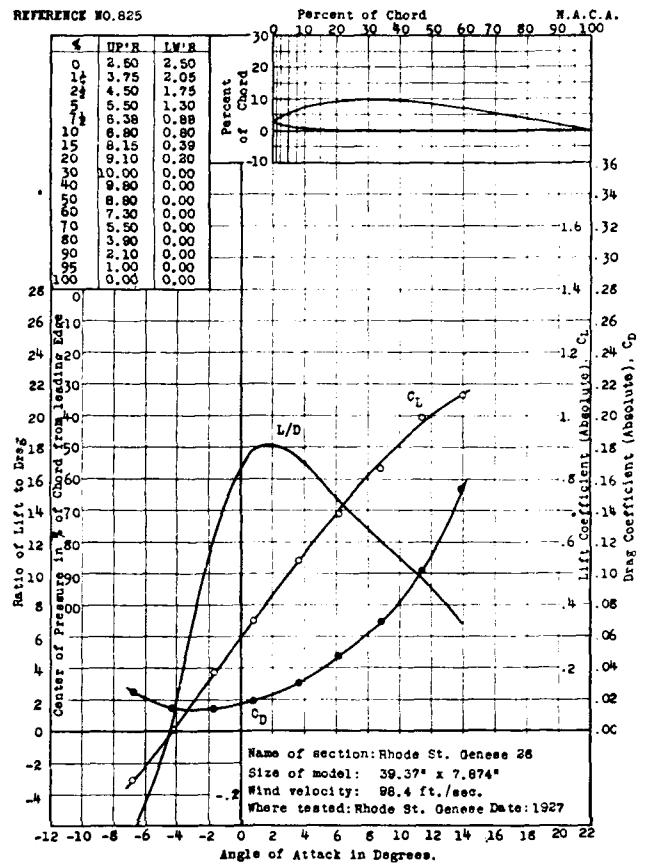
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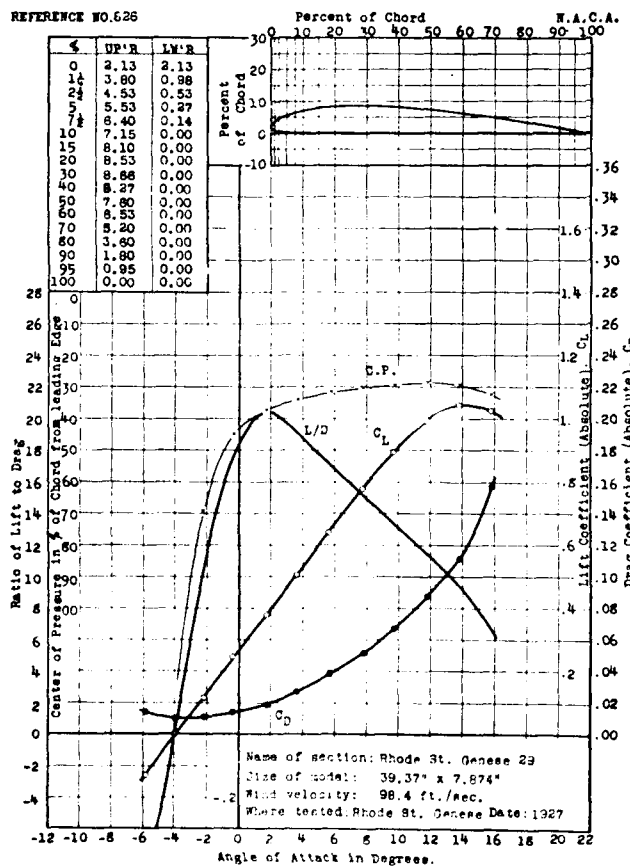
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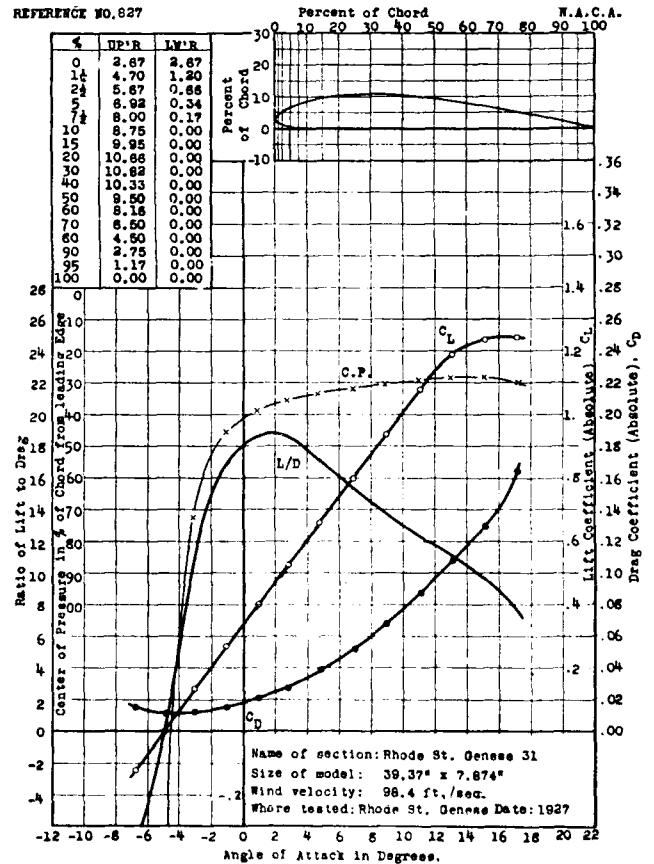
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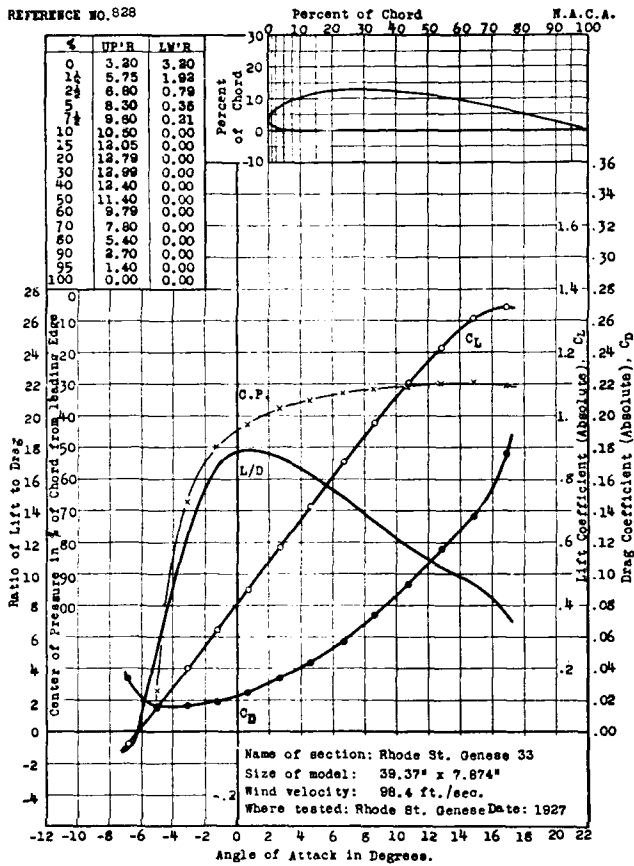
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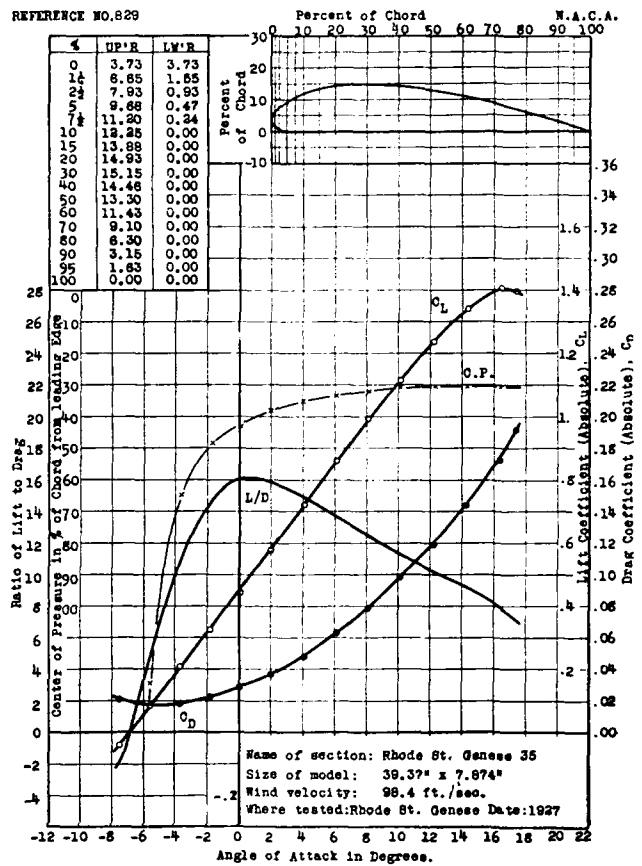
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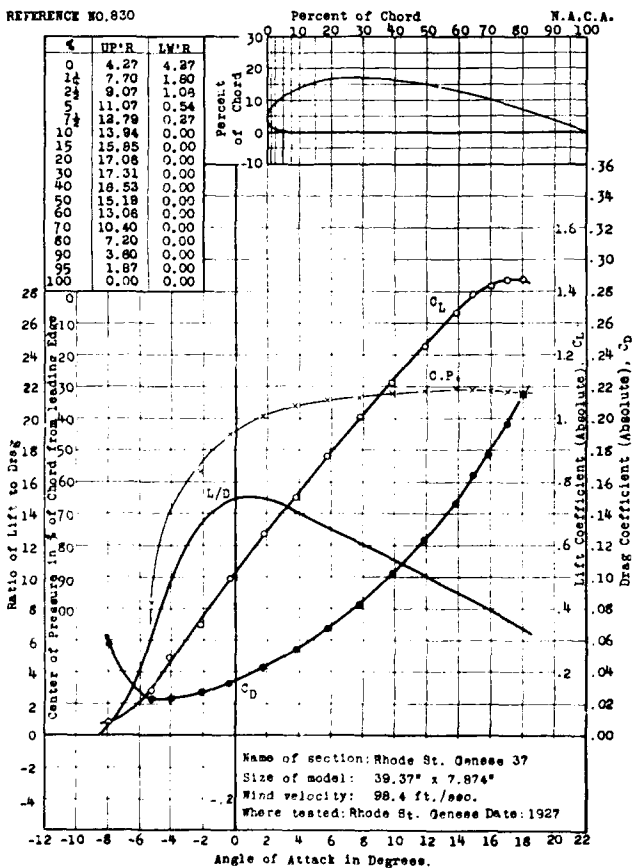
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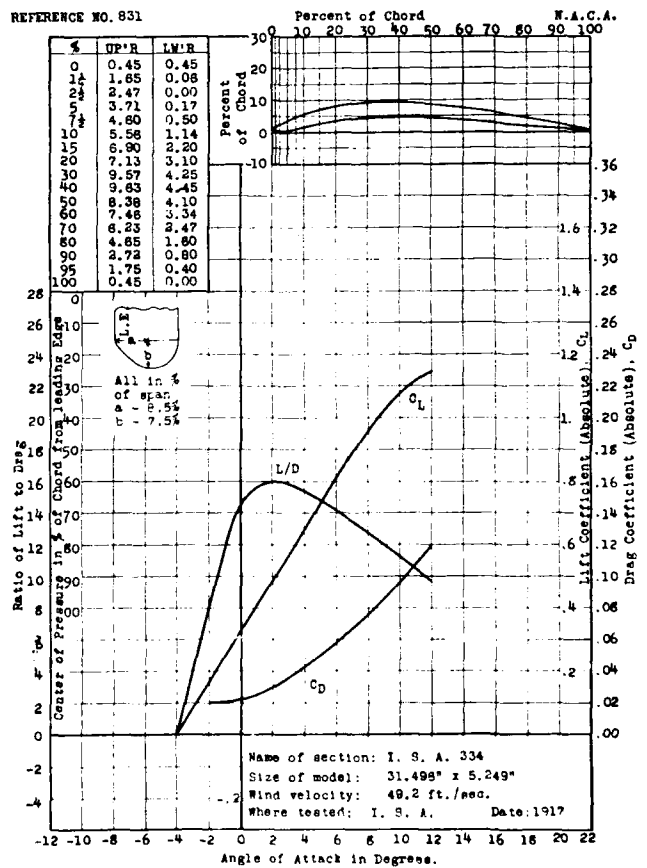
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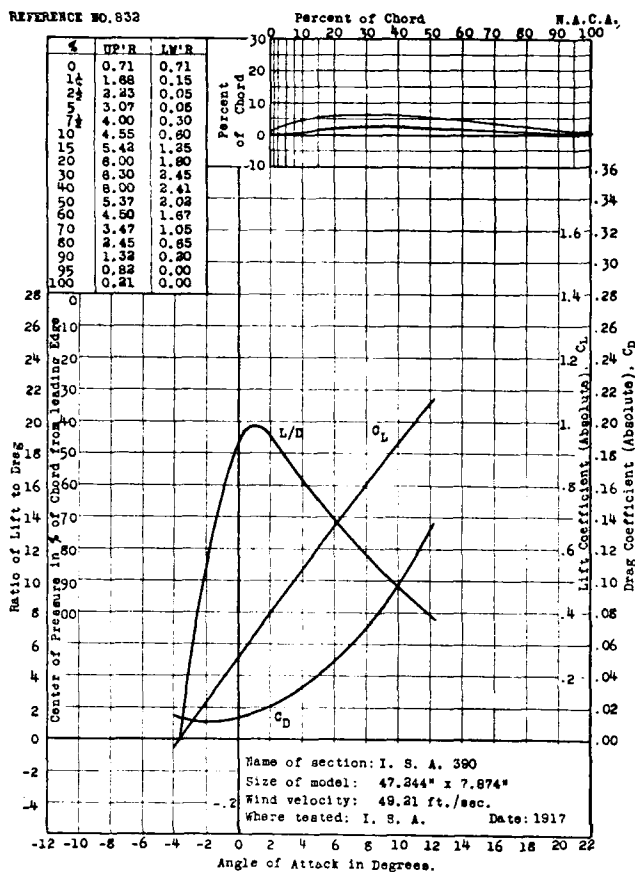
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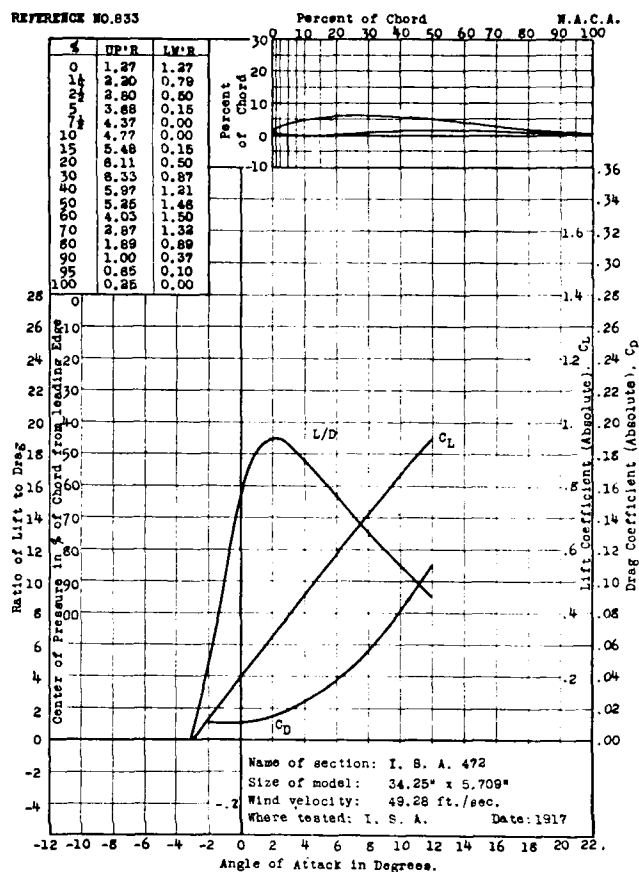
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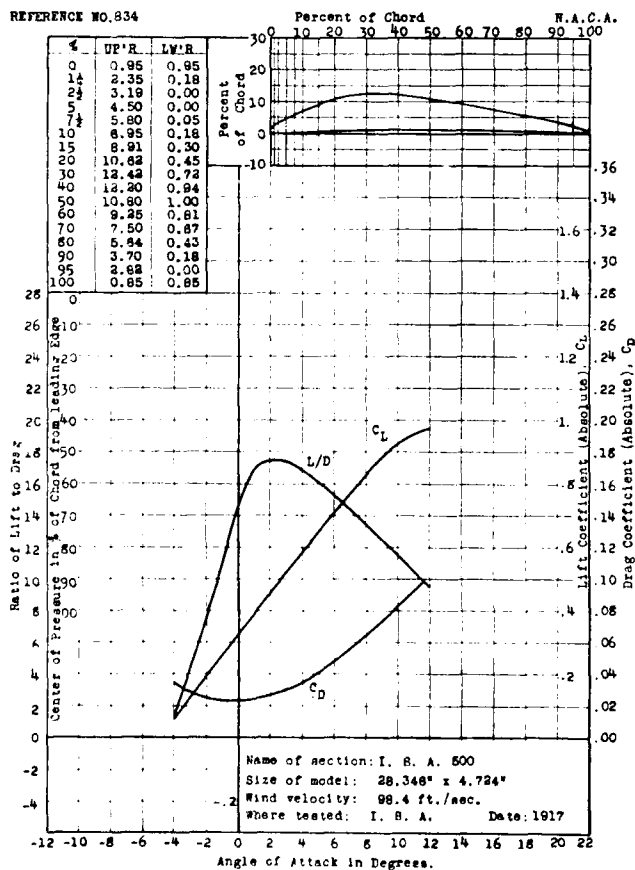
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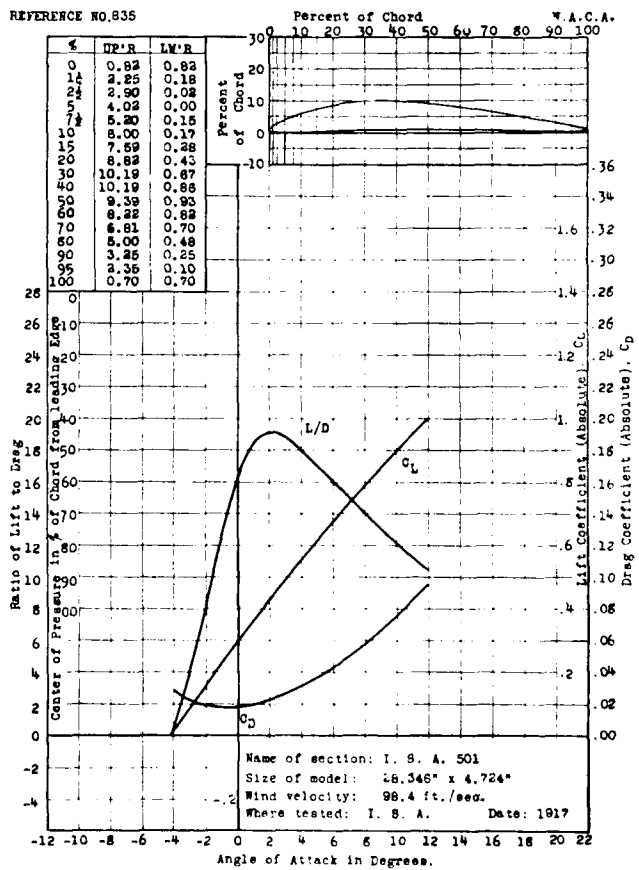
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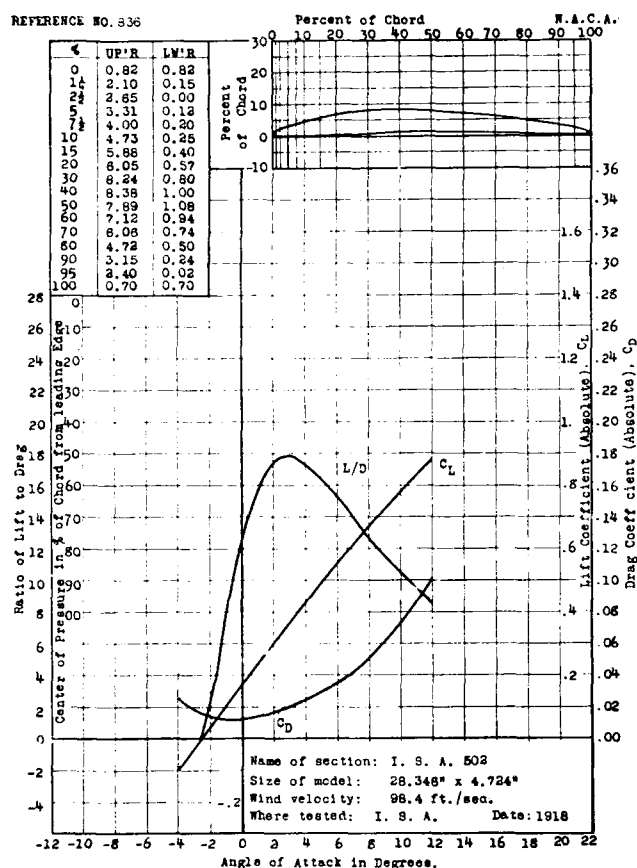
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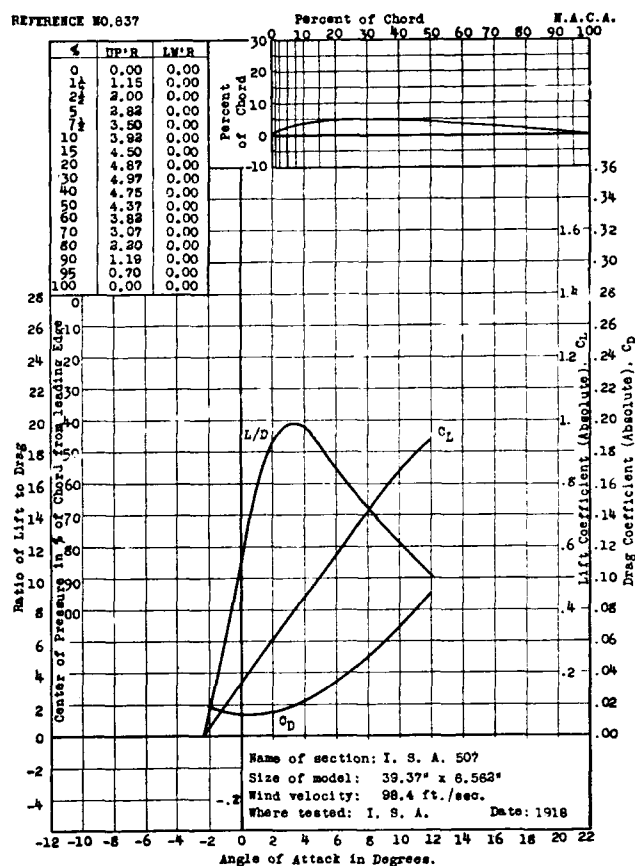
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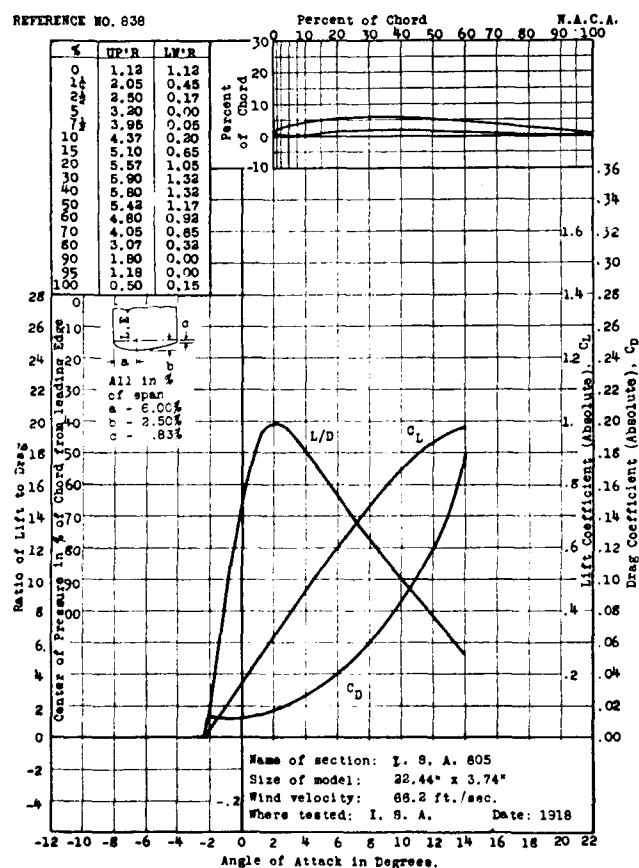
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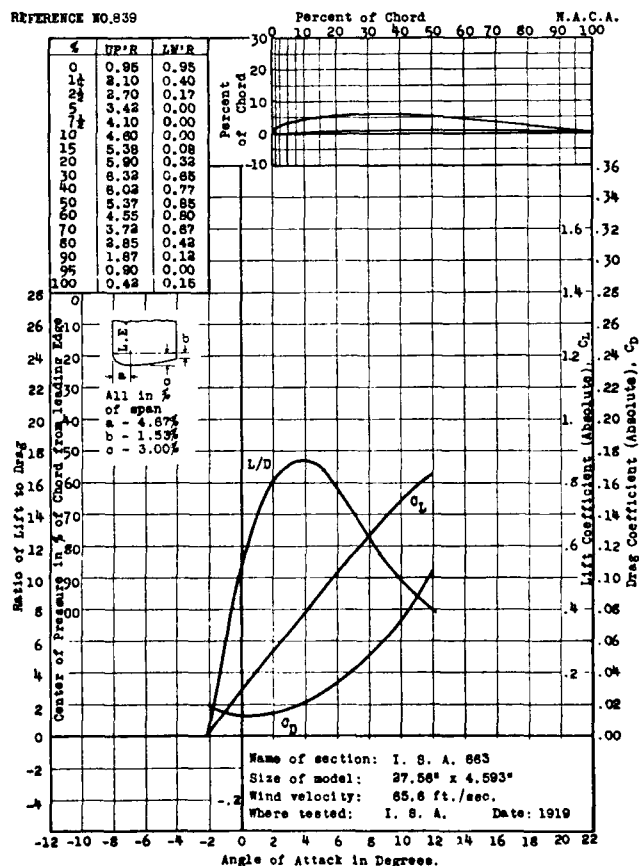
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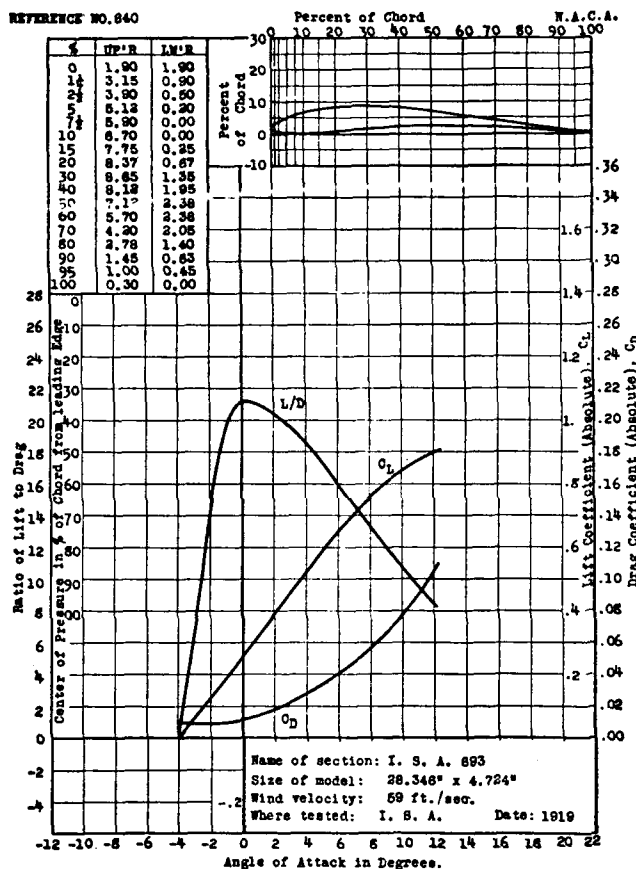
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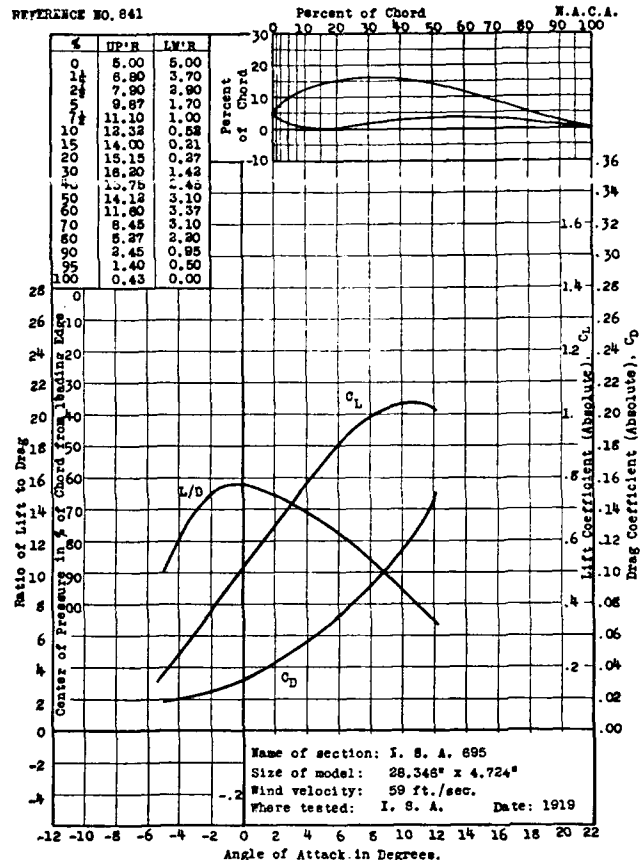
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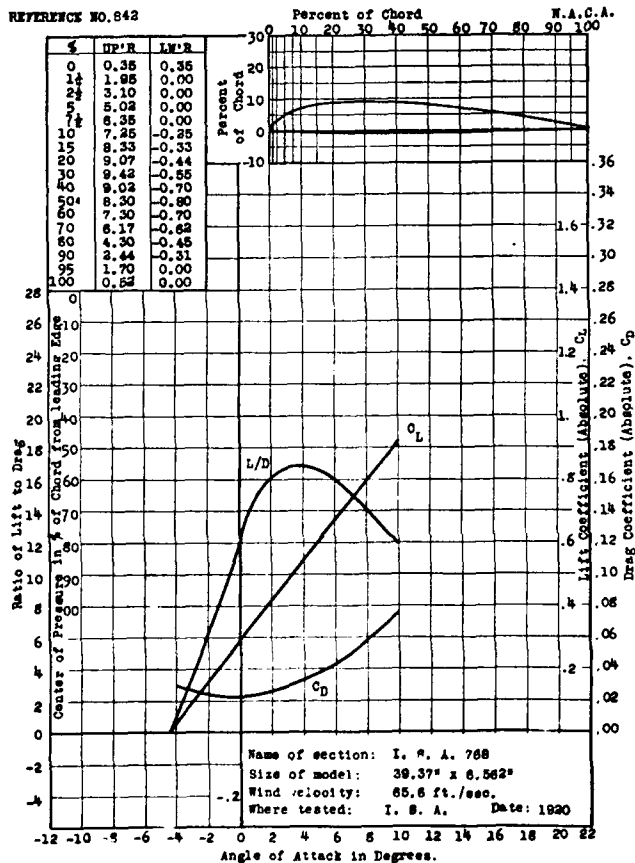
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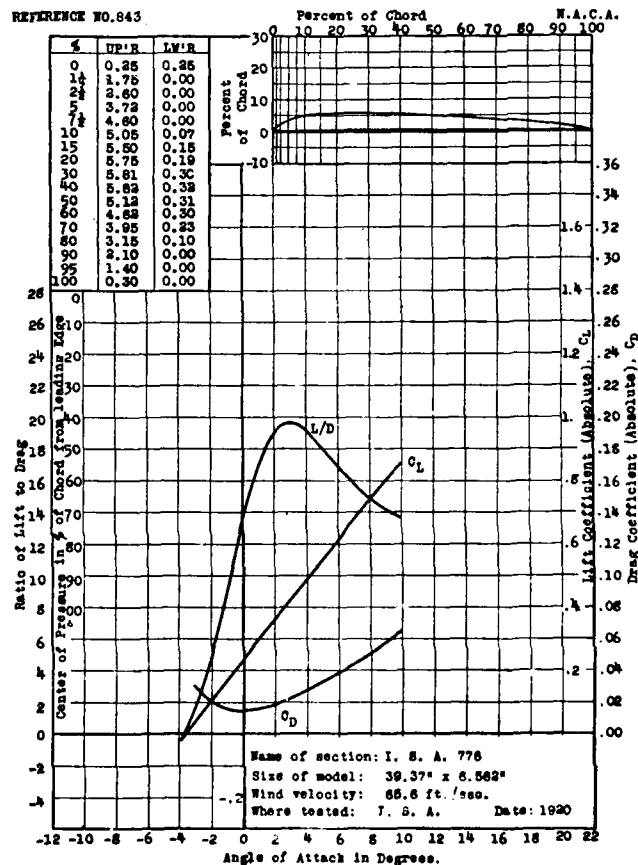
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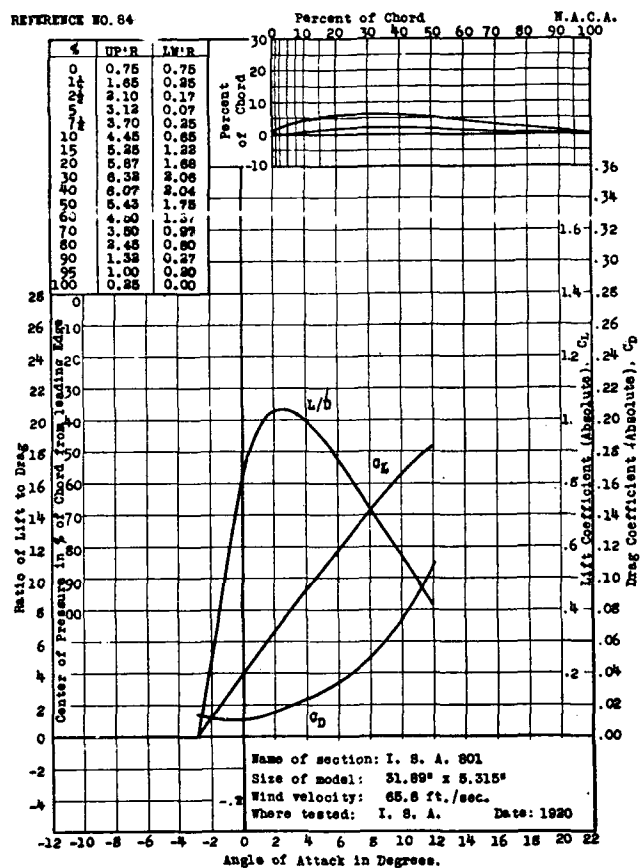
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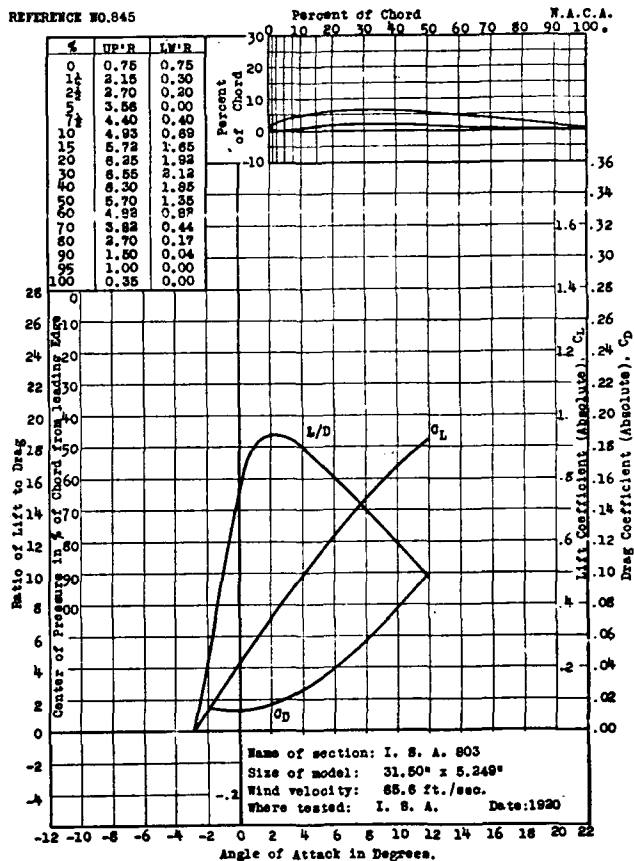
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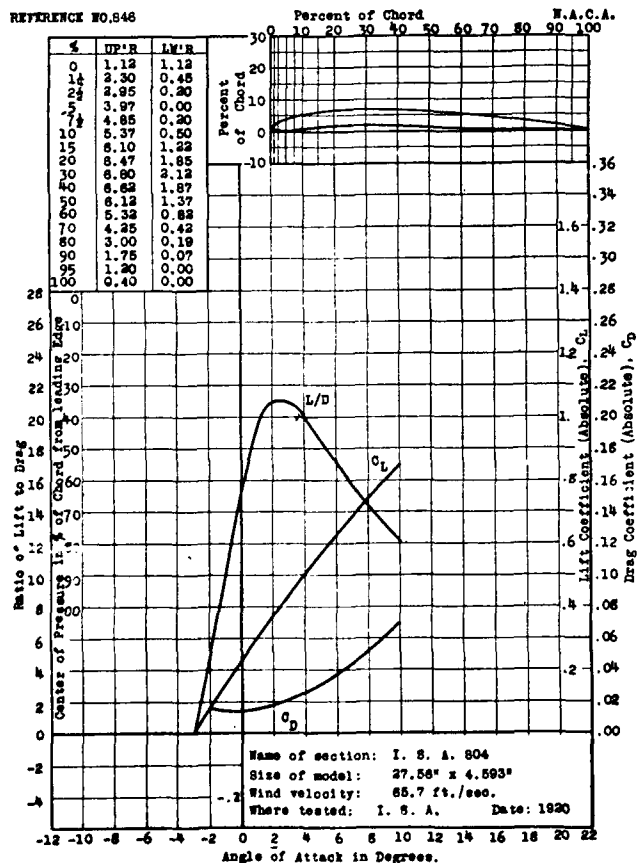
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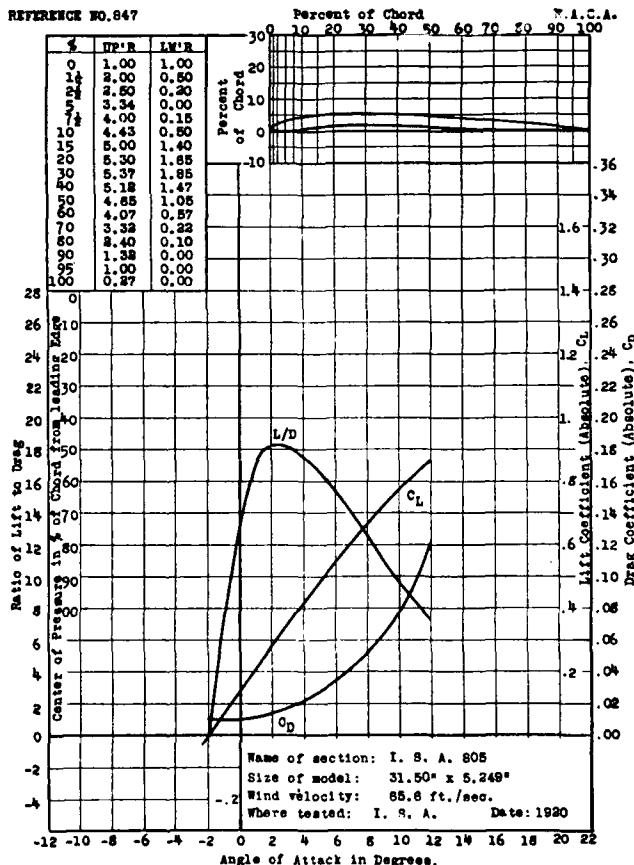
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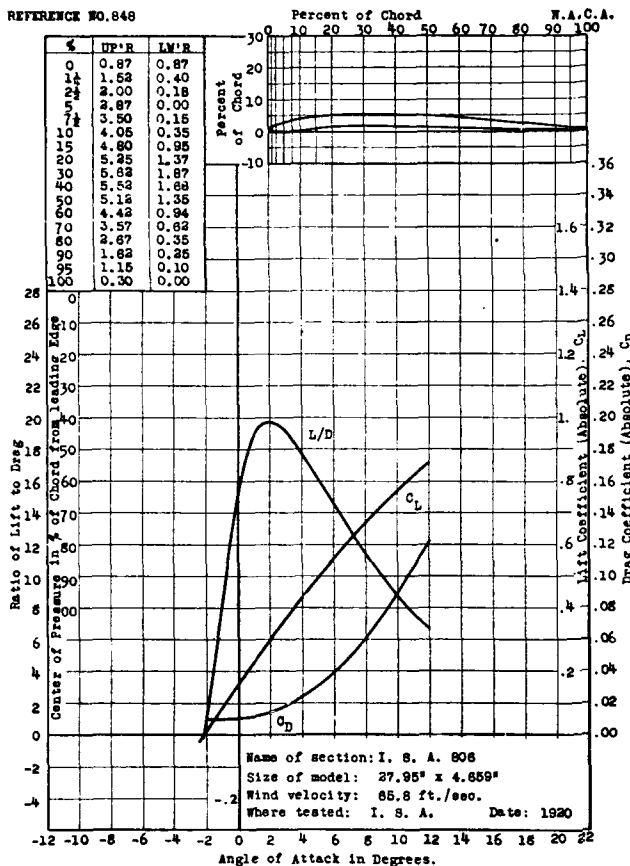


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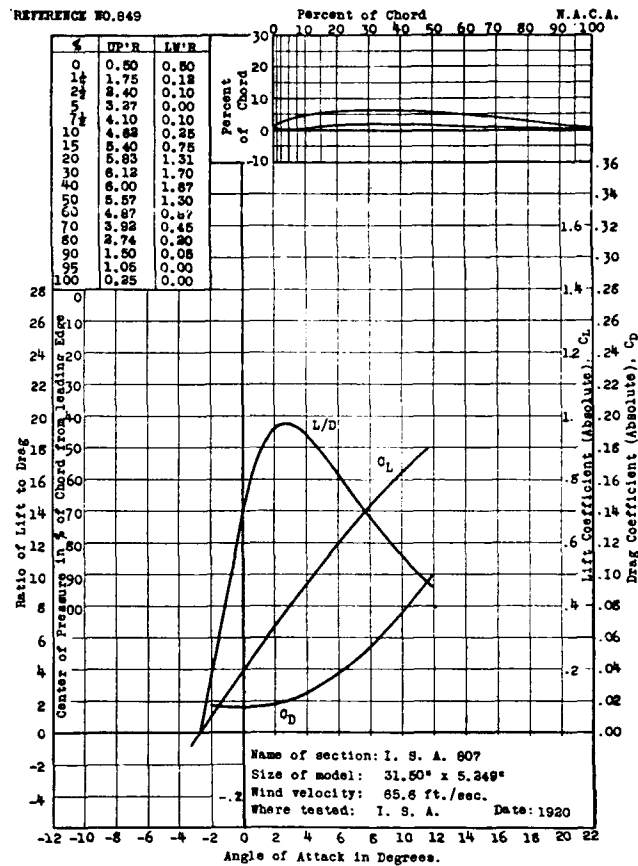




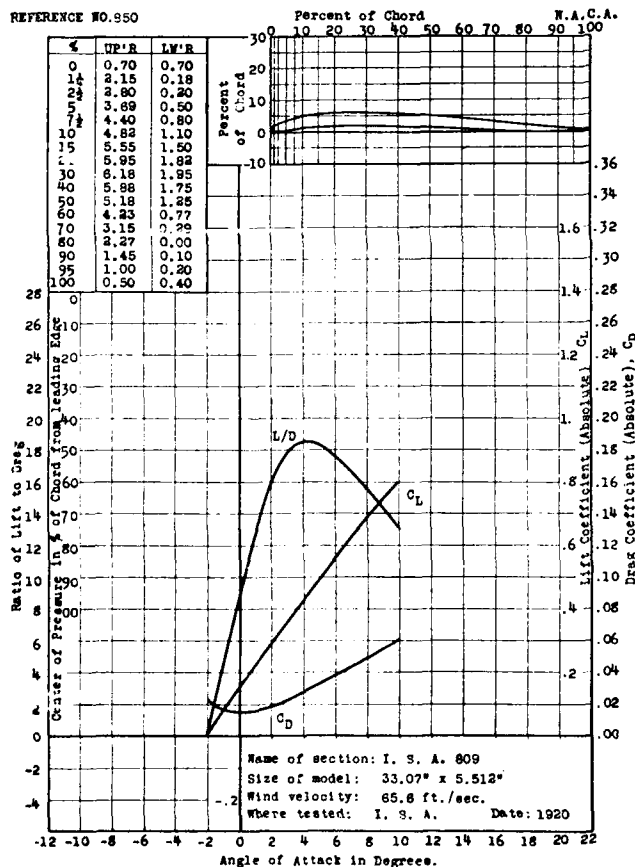
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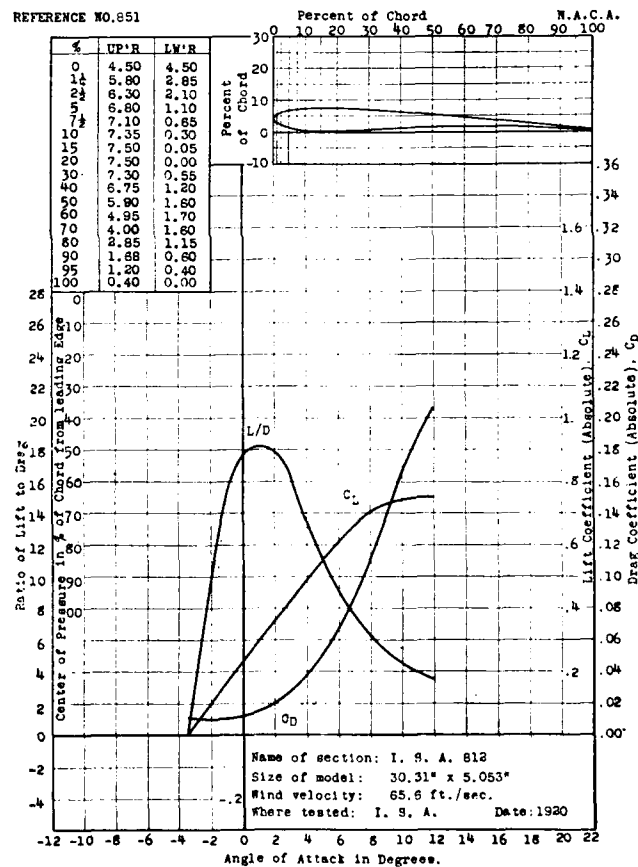
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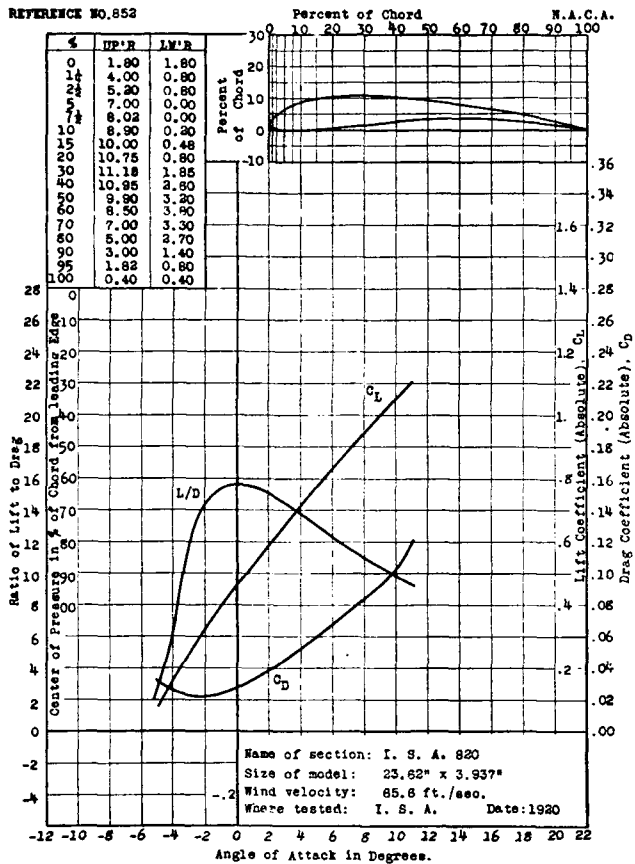
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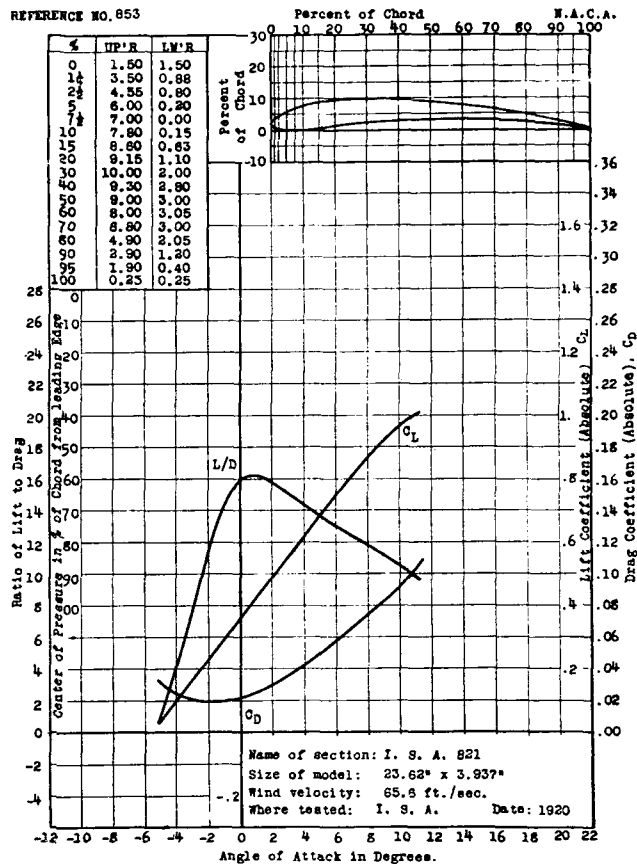
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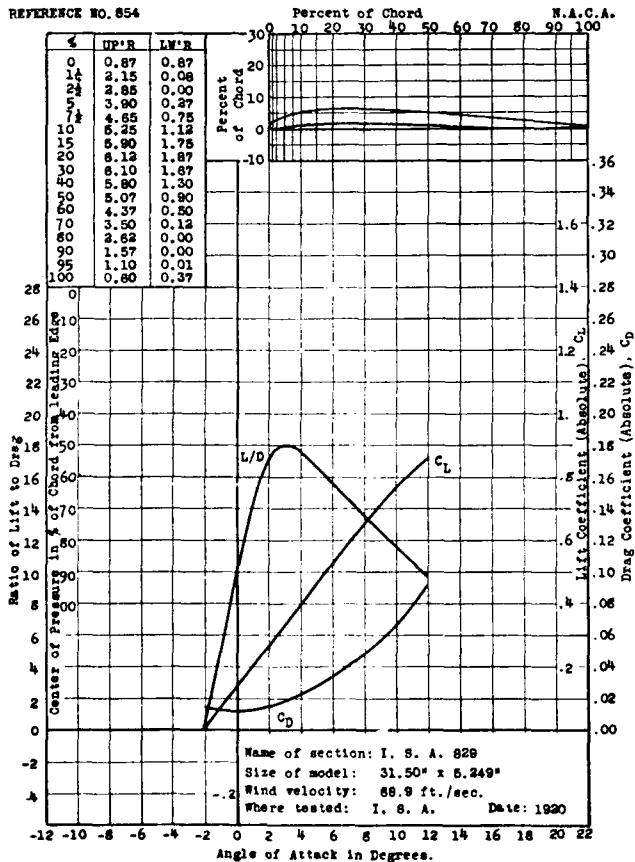
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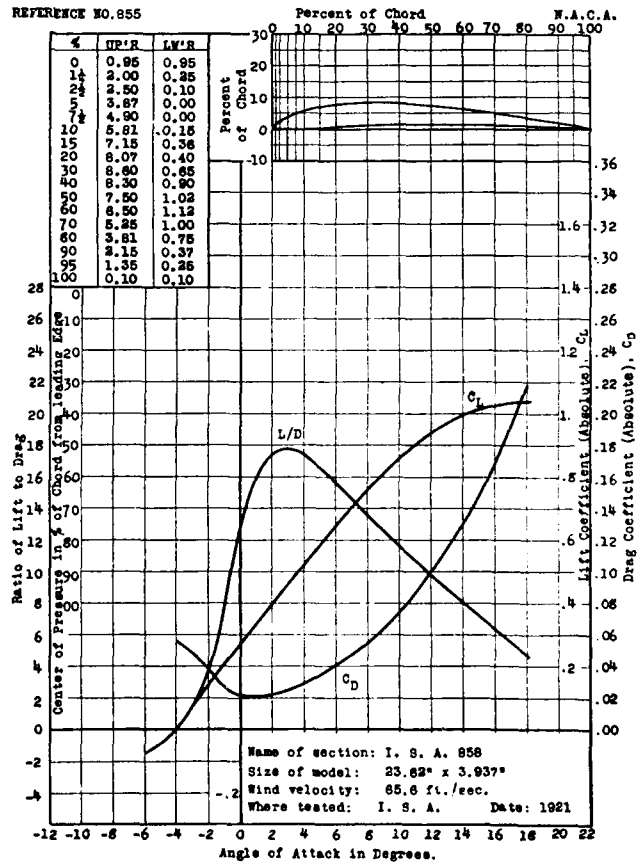
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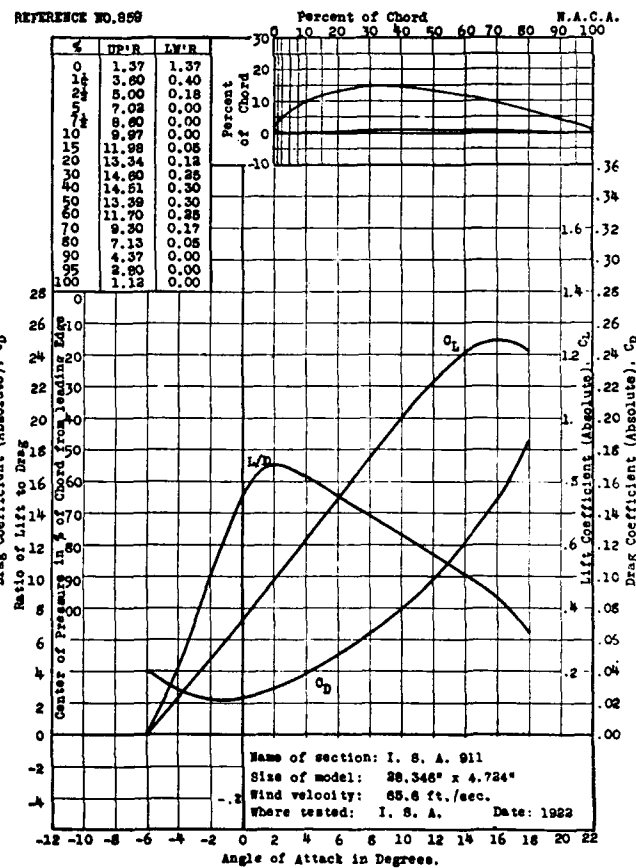
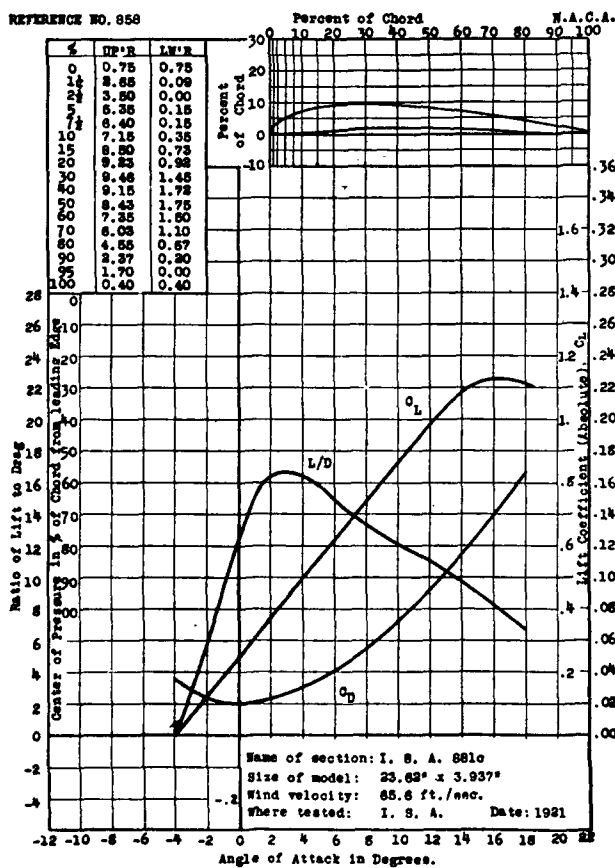
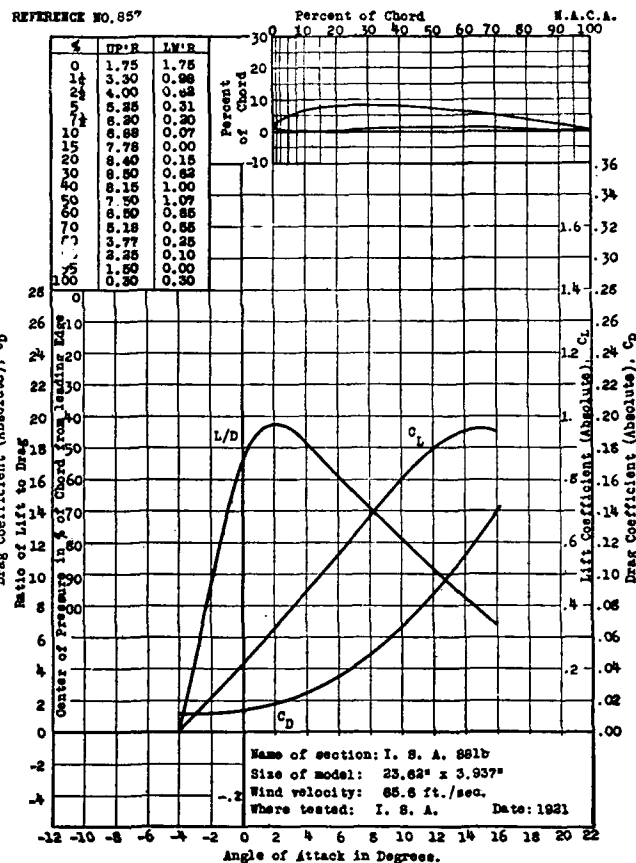
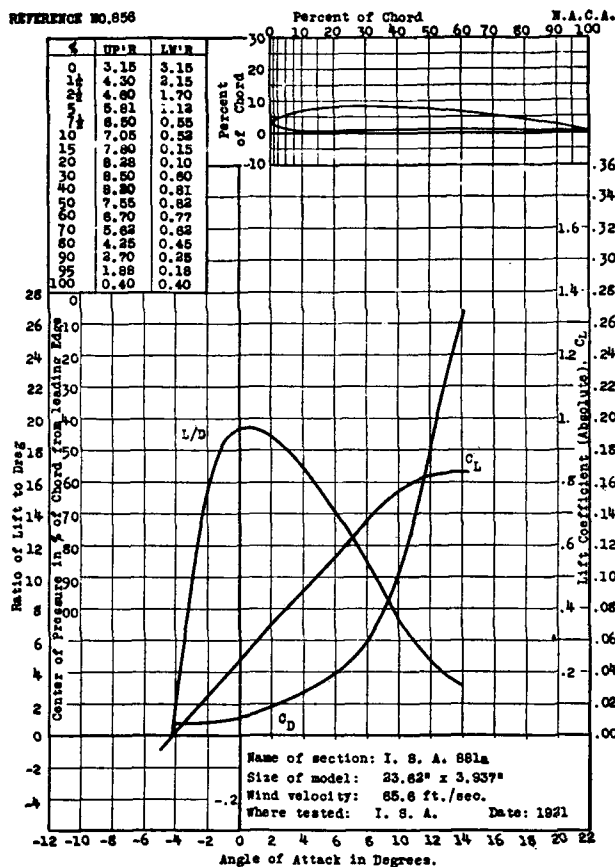


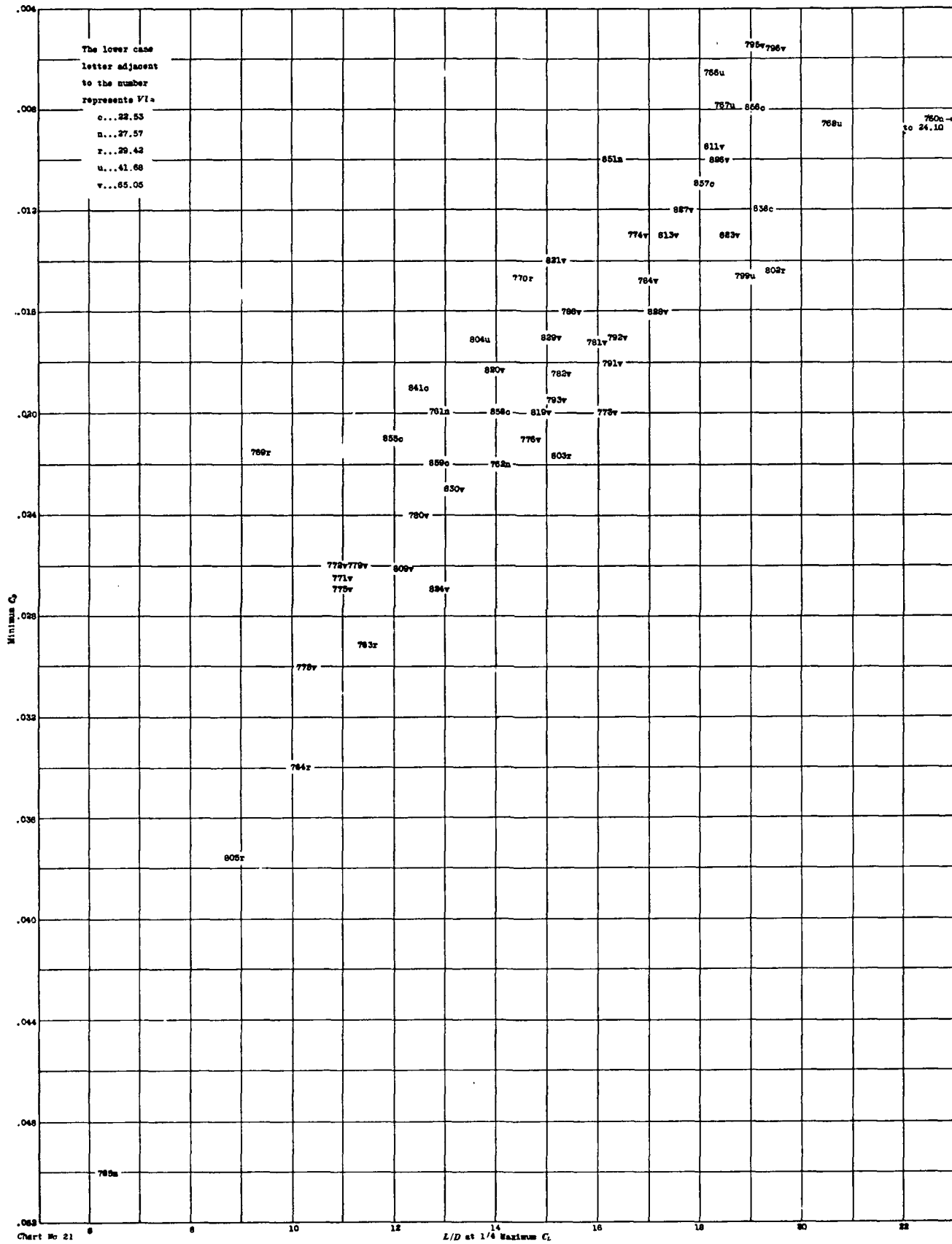
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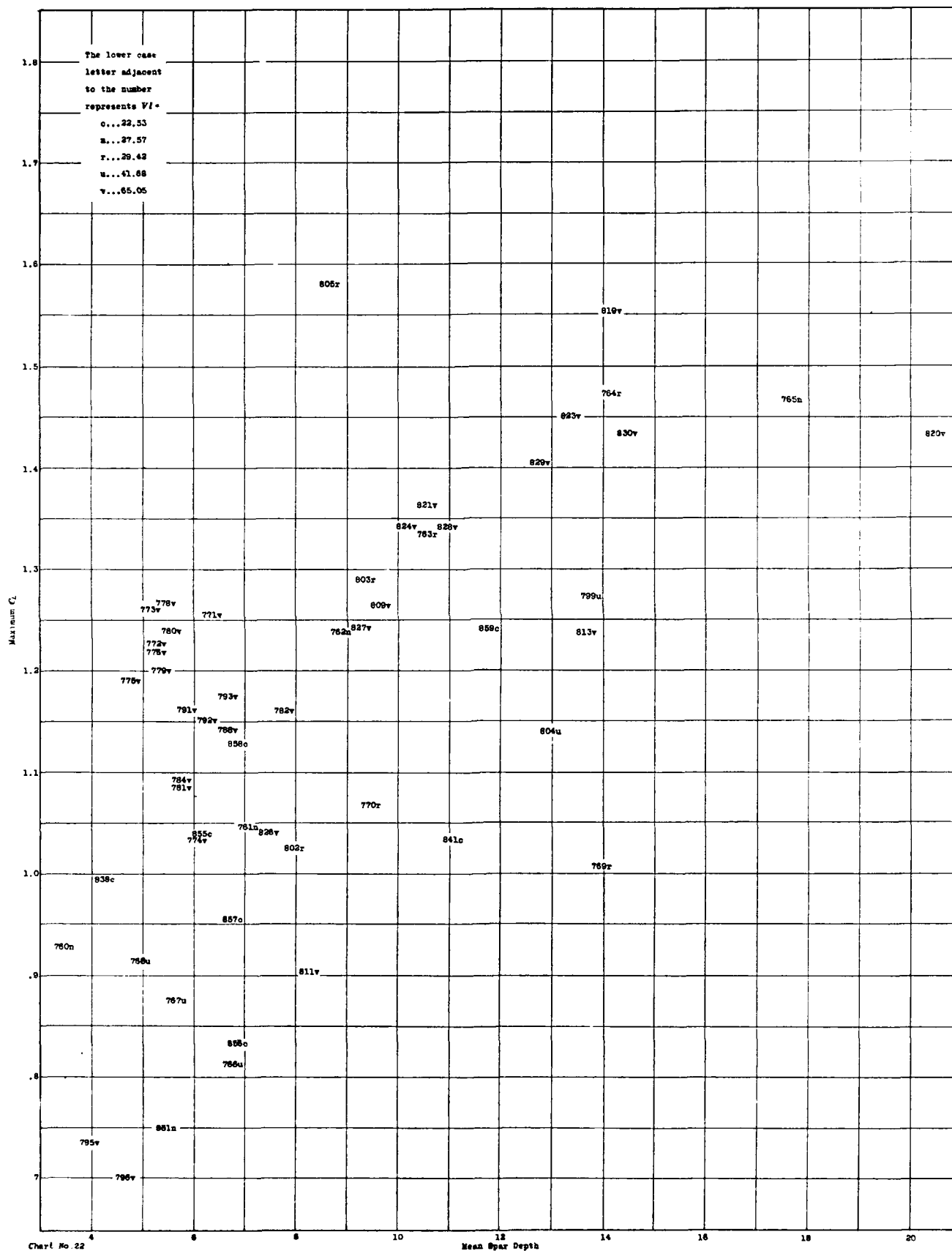


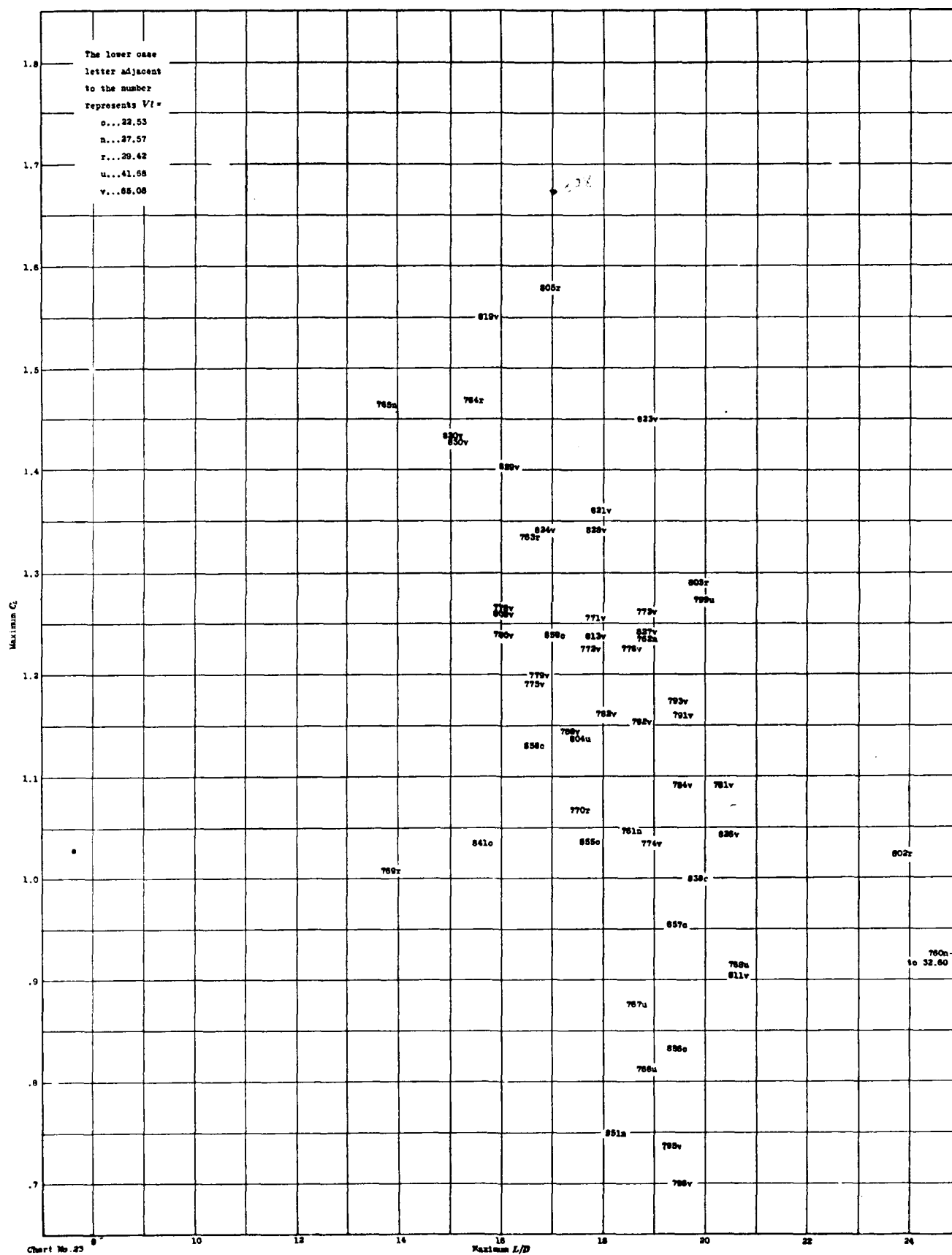
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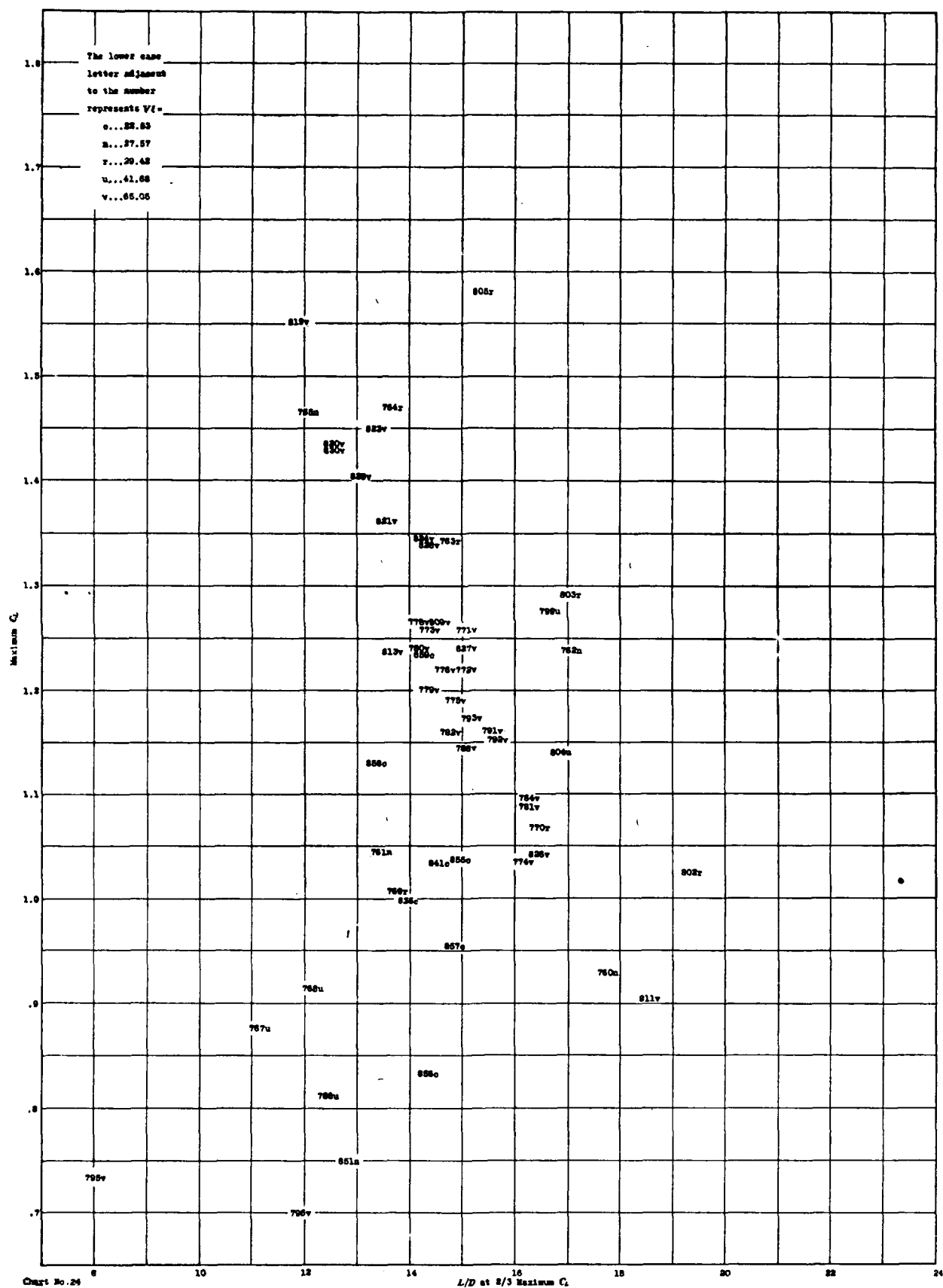






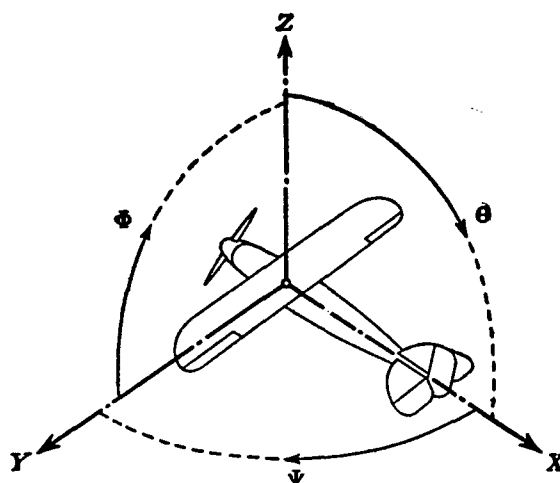






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Positive directions of axes and angles (forces and moments) are shown by arrows

Axis		Force (parallel to axis) symbol	Moment about axis			Angle		Velocities	
Designation	Sym- bol		Designa- tion	Sym- bol	Positive direction	Designa- tion	Sym- bol	Linear (compo- nent along axis)	Angular
Longitudinal....	X	X	rolling.....	L	Y → Z	roll.....	Φ	u	p
Lateral.....	Y	Y	pitching.....	M	Z → X	pitch.....	Θ	v	q
Normal.....	Z	Z	yawing.....	N	X → Y	yaw.....	Ψ	w	r

Absolute coefficients of moment

$$C_L = \frac{L}{qbS} \quad C_M = \frac{M}{qcS} \quad C_N = \frac{N}{qfS}$$

Angle of set of control surface (relative to neu-  
tral position),  $\delta$ . (Indicate surface by proper  
subscript.)

#### 4. PROPELLER SYMBOLS

$D$ , Diameter.  
 $p_e$ , Effective pitch  
 $p_m$ , Mean geometric pitch.  
 $p_s$ , Standard pitch.  
 $p_v$ , Zero thrust.  
 $p_a$ , Zero torque.  
 $p/D$ , Pitch ratio.  
 $V'$ , Inflow velocity.  
 $V_s$ , Slip stream velocity.

$T$ , Thrust.  
 $Q$ , Torque.  
 $P$ , Power.

(If "coefficients" are introduced all  
units used must be consistent.)

$\eta$ , Efficiency =  $T V/P$ .  
 $n$ , Revolutions per sec., r. p. s.  
 $N$ , Revolutions per minute., R. P. M.

$\Phi$ , Effective helix angle =  $\tan^{-1} \left( \frac{V}{2\pi r n} \right)$

#### 5. NUMERICAL RELATIONS

1 HP = 76.04 kg/m/sec. = 550 lb./ft./sec.  
1 kg/m/sec. = 0.01315 HP.  
1 mi./hr. = 0.44704 m/sec.  
1 m/sec. = 2.23693 mi./hr.

1 lb. = 0.4535924277 kg.  
1 kg = 2.2046224 lb.  
1 mi. = 1609.35 m = 5280 ft.  
1 m = 3.2808333 ft.